National Transportation Safety Board

Office of Aviation Safety Washington, DC 20594



WPR23LA098

WRECKAGE EXAMINATION SUMMARY

April 12, 2023

A. ACCIDENT

Location: Murrietta, California Date: January 27, 2023

Time: 0837pst

1637qmt

Airplane: N4758C -1979 Cessna T210N

B. WRECKAGE EXAMINATION SUMMARY

IIC Michael Hicks

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Phoenix, Arizona

Group Chair Andrew Swick

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Phoenix, Arizona

C. SUMMARY

The accident airplane, N4758C, was examined at Phoenix Air Transport, Phoenix, Arizona on 12APR2023@0730mst. The airframe examination revealed no malfunctions or anomalies that would have prevented normal operation of the airplane. Examination of the airplane's engine confirmed an engine test run which lasted less than 30 seconds. Further examination of the engine revealed impact damage to the electrical system and starter, which inhibited a conclusive determination for the loss of engine power.

D. DETAILS OF THE EXAMINATION

1.0 Airframe Examination

An airframe layout was conducted to ensure that the airframe was present in totality. The right door, the radio stack and avionics, the right wing fuel cap, as well as the nose landing gear were not present at the time of examination. The following components were separated from the fuselage to facilitate transport to a secure facility: The left and right wings, vertical and horizontal stabilizers, rudder, and the left and right elevator components.

1.1 Fuselage

The fuselage remained intact and sustained no impact damage to the bottom, left, right sides. The upper (top) fuselage area sustained impact damage as the airplane came to rest inverted. Each the plexi-glass windows were present, but the windscreen was grossly missing. The left and right main landing gear were present and found in the extended and locked position, while the nose landing gear had sustained impact damage and was departed the airplane.

1.2 Empennage

Examination of post-accident photographs revealed that the empennage remained attached to the fuselage but was separated from the fuselage for recovery transport. The vertical stabilizer and rudder remained attached to each other via their respective hinges. They both sustained impact damage and when viewed from the nose of the airplane, were bent to the left, emanating from the top of each structure. The horizontal stabilizer sustained minor impact damage to the left side. The left and right elevator were present, intact, and sustained minor impact damage to the surfaces. The elevator trim tab remained affixed to the right elevator hinge and remained intact.



Figure 1: View of damage to the empennage (Courtesy of NTSB)

1.3 Wings

The right wing sustained impact damage to the leading edge and remained attached via the spar, which was fractured near the middle of the wing. The wing tank was breached, and the vented fuel cap was not present. The outboard section sustained crushing signatures emanating from the outboard wing tip, to about the middle of the wing. The aileron and flap remained attached to their respective hinges and sustained impact damage.

The left wing sustained most of the impact damage to the leading edge and the outboard sections. The center of the wing sustained crushing signatures from the forward to aft surface, and the wing tank was breached. The vented fuel cap was present and unremarkable. The outboard section sustained crushing signatures emanating from the wing tip and about 10 percent of the outboard section of the wing, including the outboard section of the aileron which remained attached to its respective hinges, had folded underneath the wing. The flap remained attached to its respective hinges and sustained minor impact damage.



Figure 2: View of damage to right wing (Courtesy of NTSB)



Figure 3: View of damage to left wing (Courtesy of NTSB)

1.4 Landing Gear

The left and right main landing gear were down and locked and appeared unremarkable. The nose gear separated from the lower fuselage and appeared down and locked.

1.5 Cockpit

The instrument panel was present and did not sustain impact damage. However, the HSI and the attitude indicator were missing from the pilot side (left) of the instrument panel, and the radio stack was missing from the center of the instrument panel. There were two LED screens, both about 2 x 2 inches, that were present and not damaged. The manifold pressure and fuel flow gage indicated zero for both. The battery amperes gage indicated zero amps. The oil pressure gage indicated zero psi. The cylinder head temperature gage indicated zero degrees. The engine RPM gage indicated zero RPM, and the tachometer indicated 1,136.63 hours. The altimeter indicated 1,475 ft and the Khlosman window indicated 30.24. The VSI indicated a 150 ft climb per minute. The airspeed indicator indicated zero knots, and the turn and slip indicator indicated level, with ½ ball left of the trim. The defrost, cabin heat, cabin air, and aux cabin air were all in the Off position.

The landing gear handle was in the gear down and locked position. The cockpit propeller lever was in the maximum RPM position. The mixture was in the full rich position. The throttle was in the full forward position. The CDI and OBS indicated a course to flag 050 degrees, with the NAV flag present. The elevator trim indicated a slightly nose up position, and the rudder trim indicated a neutral position. The cowl flap lever was in the closed position.

The NAV lights, Taxi lighting and Pitot heat switches were in the Off position. The strobe and beacon lighting were in the On position. All circuit breakers were in, the master switch (Alt and Battery) and aux pump were in the off position and the ignition switch appeared to be in the Off position. The Aspen glass cockpit panel that was missing, from the pilot side, had a switch on the left side of the cockpit that indicated that it was in the On position. The Autopilot switch indicated being in the On position, as did the auto-trim switch. The fuel tank selector handle was in the Off position. The left and right tank fuel quality indicators both indicated zero fuel.

1.6 Fuel/Fuel System

The left and right fuel tank were breached. The left fuel cap was present, but the right fuel cap was not present. The wings were detached from the fuselage, and the fuel pick-up and return lines were clear and uninhibited. The fuel lines above the left and right fuselage doors, specifically, the fuel pick-up lines to the fuel selector were connected to an auxiliary external fuel source mounted above the cockpit in preparation for the engine test run. Continuity was observed from the left and right inboard pick-up lines to the fuel selector and the fuel selector was moved from the Off, Left and Right tank positions without impedance. Additionally, continuity was observed from the fuel selector to the fuel manifold inlet. The fuel injectors were removed and observed as being clear and free of impedance. The auxiliary fuel pump was turned on, and fuel flow was observed through the manifold at every cylinder. The engine driven fuel pump was removed from the crank case after the test run and disassembled. No anomalies or failures were observed pertaining to internal components of the vacuum type, engine driven fuel pump.

2.0 Engine Examination



Figure 4: View of examined engine data plate (Courtesy of NTSB)

The airplane and separated parts resulting from the accident sequence were displayed on the facility floor. The engine and cowling remained partially attached to the fuselage as result of impact damage to the engine mounts. The engine was pitched down from its normal position about 15-degrees and the upper cowling was separated from the firewall attachment. The lower cowling sustained impact damage. The right half of the upper cowling sustained impact damage. The upper cowling was removed from the engine compartment.

The magneto p-leads had pulled from the magneto attaching hardware. A large wire had pulled from the alternator control relay located on the firewall, and the alternator wire hardware was loose.

Engine control continuity was established from the cockpit. The throttle, mixture and propeller control cables took effort to move resulting from the damaged engine resting position, and the movement was accomplished from stop to stop. The engine was repositioned to near its normal position on a mounting carriage and engine control continuity improved.

The right induction tubes inlet pressure sensor separated from the inlet tube fittings.

The oil dip stick read about 7 quarts. The oil was clear.

Manifold valve was disassembled, and fuel was present in its cavity. The diaphragm was undamaged, and the inlet screen was clear of debris.

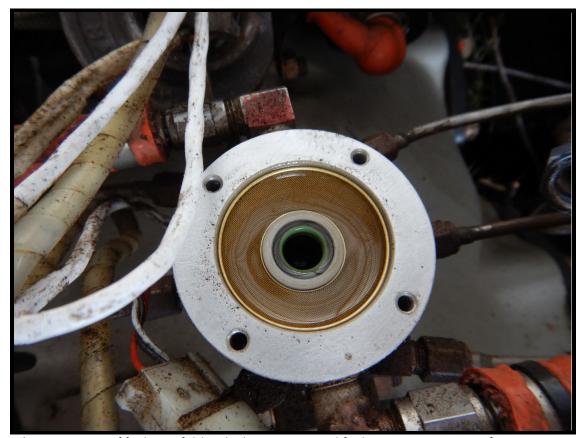


Figure 5: View of fuel manifold with clean screen and fuel present (Courtesy of NTSB)

The upper spark plugs were removed, and their electrode areas had a dark residue, and the electrode areas had a (normal wear) wear condition consistent with the Champion Aviation Service Manual. The spark plugs were reinstalled.

During the first engine start attempt the engine would turnover. Relays were audibly heard during the preparation of the engine start. It was determined that wire damage near the battery prevented power to supply the starter relay. Multiple circuit breakers near the battery position were damaged and replaced. An auxiliary power source was used to connect power to the starter relay, and the starter operated normally from the switch. The engine started successfully but the engine ran with some hesitation and quit less than 30-seconds into the run. During the next few attempts to start the engine the starter would spin up and not engage for engine rotation.

The lower spark plugs were removed, and their electrode areas had a dark residue, and the electrode areas had a (normal-worn out) wear condition consistent with the Champion Aviation Service Manual. The No. 2 lower spark plug electrode area was oil soaked.

The cylinders were observed with a borescope and each of the piston heads and combustion areas revealed light combustion deposits. The cylinder valves in each of the cylinders showed normal operating signatures.

The magnetos were removed from the engine and the drive shafts were rotated by hand. The impulse coupling worked normally, and spark was produced at each ignition lead end.

The manifold inlet line was removed, and the auxiliary pump was switched to the On position. Subsequently, the fuel selector was positioned in all three positions and fuel flowed freely from the separated line.

The fuel nozzles were removed from the cylinders and were clear of obstructions.



Figure 6: Image of bottom spark plugs and fuel injectors (Courtesy of NTSB)

The turbocharger was undamaged and the wastegate valve was found in the open position.

The rocker covers were removed from their respective cylinders and the engine was rotated through by hand, and each rocker showing equal travel on each cylinder intake and exhaust.

3.0 Propeller Examination

The propeller assembly remained attached to the engine and had impact damage to each of its three blades. The blades were bent aft and had multidirectional scuffing. Some material was missing from two of the blades. The spinner had crushing on one side. The propeller assembly was removed and replaced with a 2-bladed propeller in preparation for an engine test run.



Figure 7: View of the damage to propeller blades (Courtesy of NTSB)

4.0 Aircraft Logbook Information

Engine

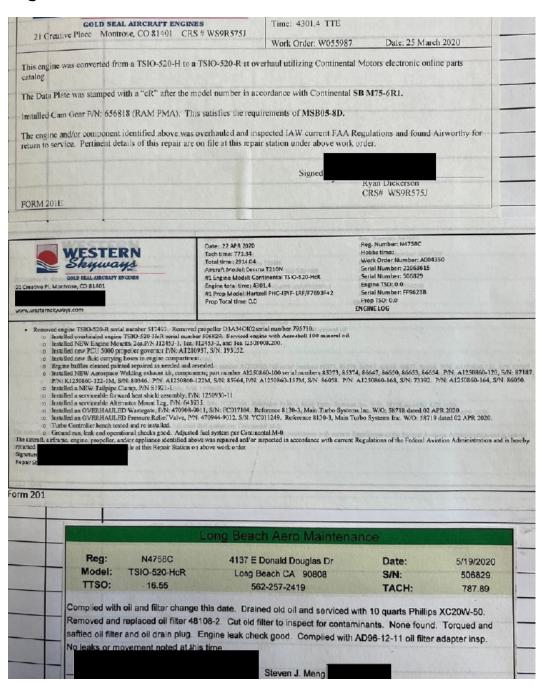


Figure 8: View of engine logbook (Courtesy of PIC)

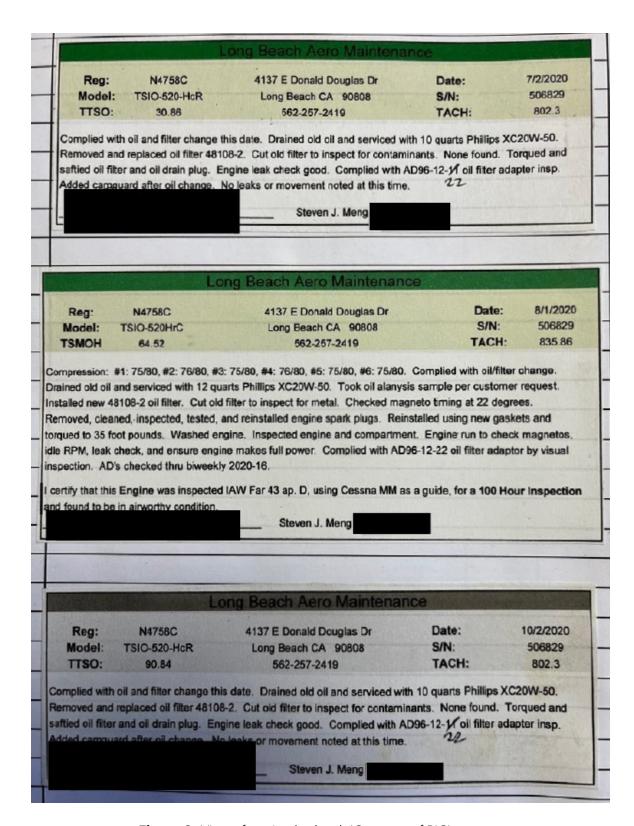


Figure 9: View of engine logbook (Courtesy of PIC)

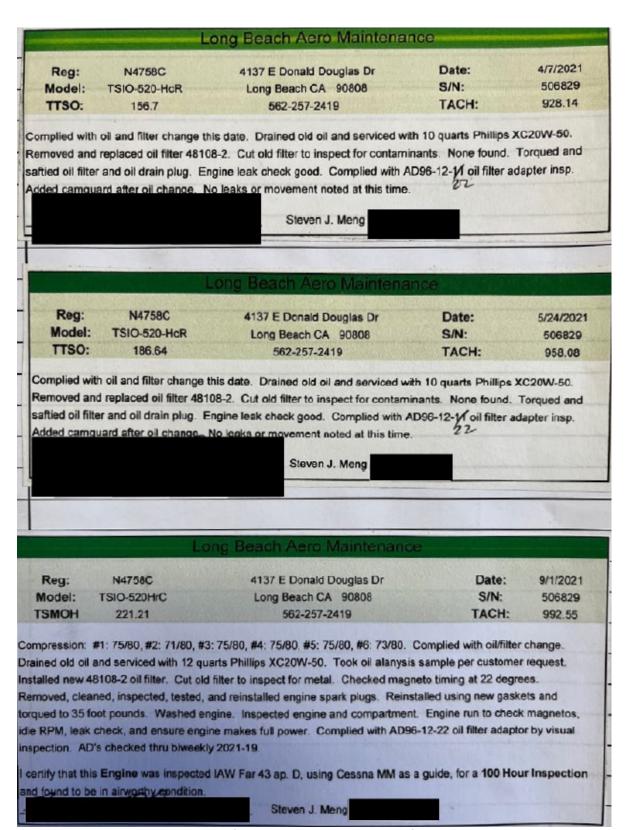


Figure 10: View of engine logbook (Courtesy of PIC)

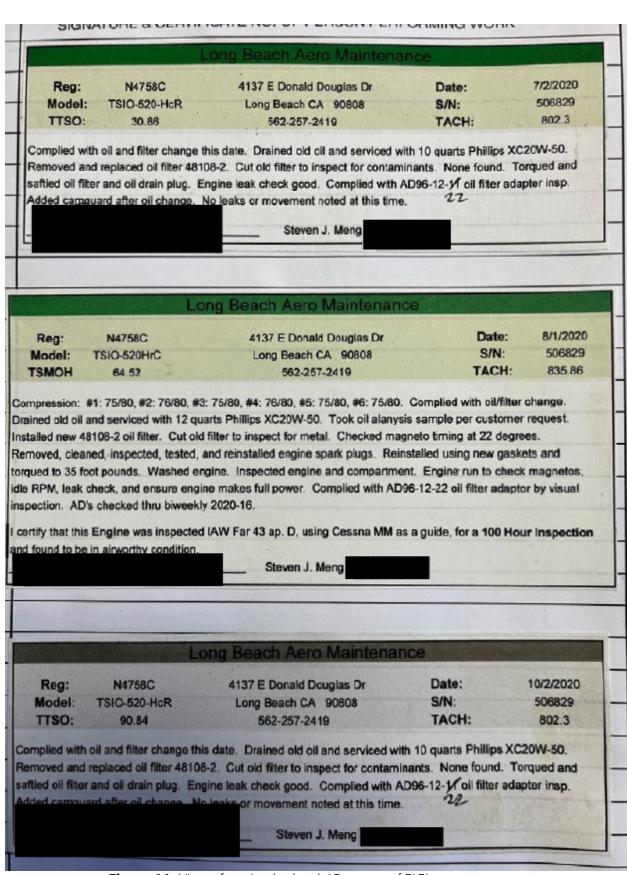


Figure 11: View of engine logbook (Courtesy of PIC)

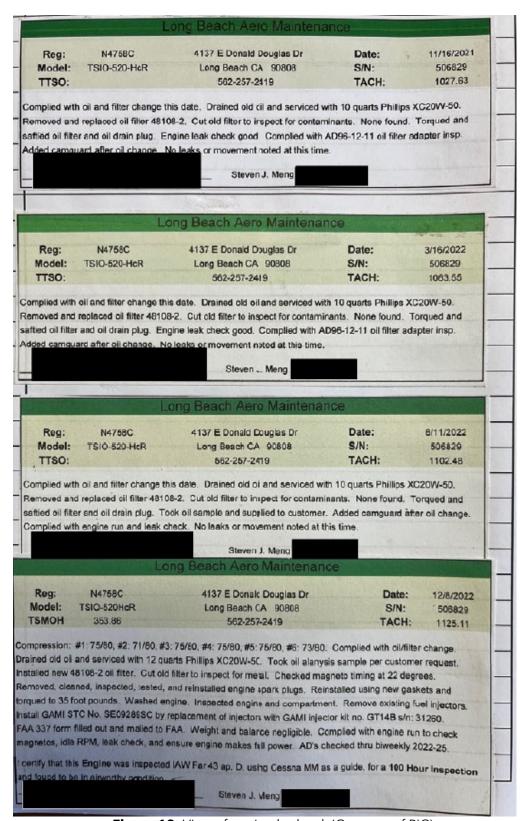


Figure 12: View of engine logbook (Courtesy of PIC)

Propeller Long Beach Aero Maintenance Reg: N4758C Date: 8/1/2020 Model: PHC-J3YF-1RF/F7693F+2 S/N: FF9623B TTSN: 64.52 TACH: 835 86 Cleaned, inspected, dressed, and repainted propeller as necessary. AD's checked thru biweekly 2020-16. I certify that this Propeller was inspected IAW Far 43 Ap. D, using Cessna MM as a guide for a 100 Hour Inspection and found to be in airworthy condition. Steven J. Meng Reg: N4758C Date: 9/1/2021 Model: PHC-J3YF-1RF/F7693F+2 S/N: FF9623B TACH: 992.55 TTSN: 221.21 Cleaned, inspected, dressed, and repainted propeller as necessary. AD's checked thru biweekly 2021-19. I certify that this Propeller was inspected IAW Far 43 Ap. D, using Cessna MM as a guide for a 100 Hour Inspection Steven J. Meng

Figure 13: View of propeller Logbook (Courtesy of PIC)

Airframe

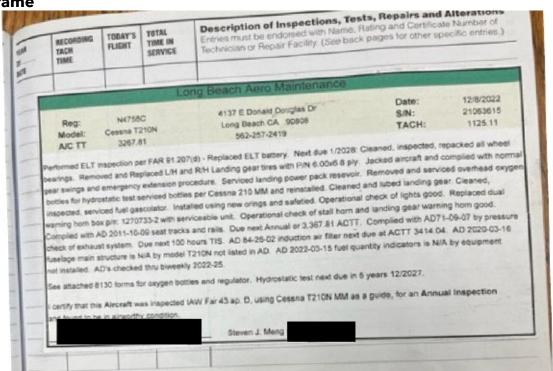


Figure 14: View of airframe logbook entry (Courtesy of PIC)

Submitted by:

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