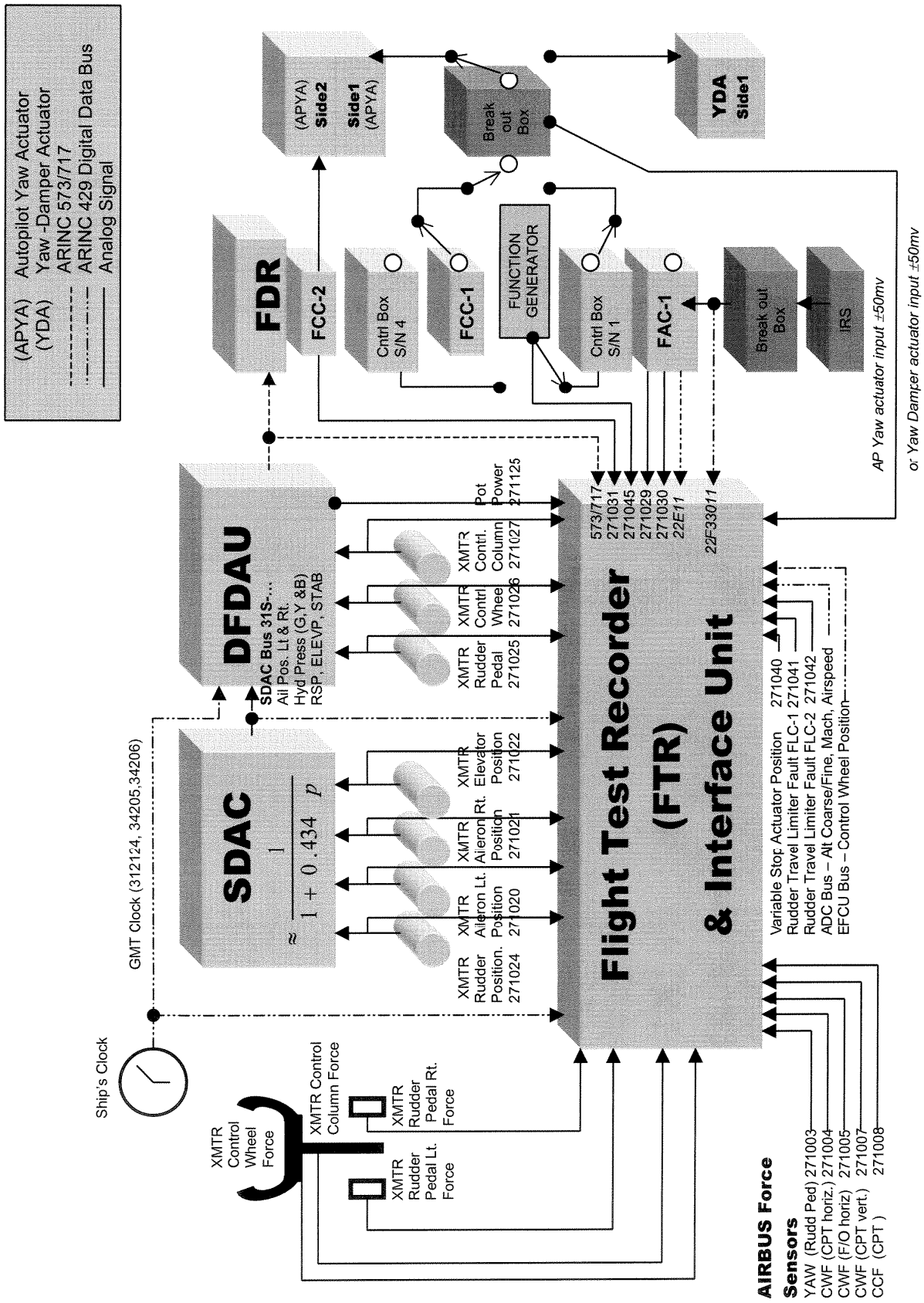
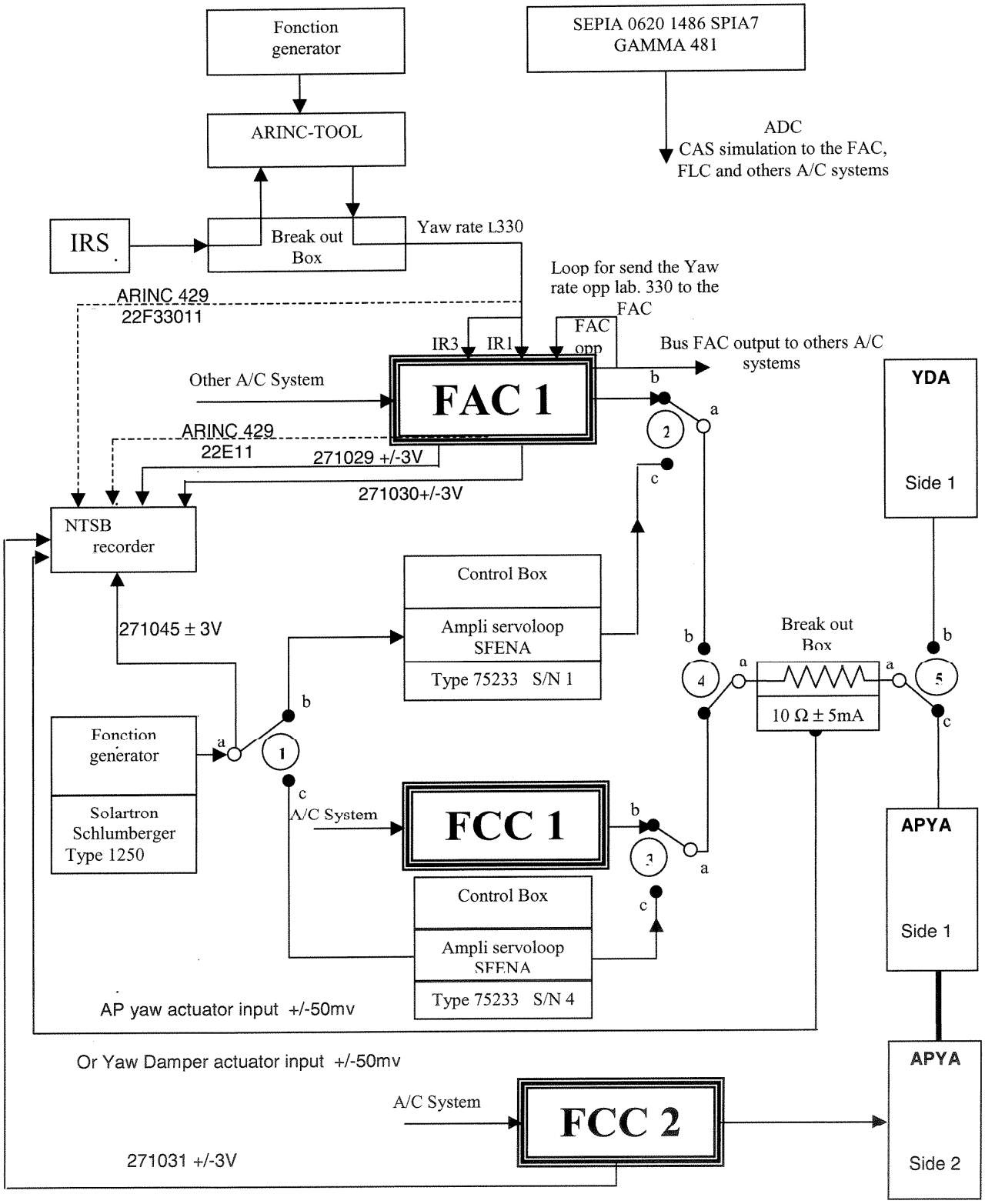
Appendix B

To

ADDENDUM NUMBER 4 TO THE SYSTEMS GROUP CHAIRMAN'S FACTUAL REPORT OF INVESTIGATION - A300-600 GROUND TEST


(APYA) Autopilot Yaw Actuator
(YDA) Yaw -Damper Actuator
ARINC 573/717
ARINC 429 Digital Data Bus
Analog Signal
FDR
(APYA) Side2
Side1 (APYA)
YDA Side1
Break out Box
FCC-2
FCC-1
Cntrl Box S/N 4
FUNCTION GENERATOR
Cntrl Box S/N 1
FAC-1
Break out Box
IRS
DFDAU
SDAC Bus 31S-...
Ail Pos. Lt & Rt.
Hyd Press (G,Y &B)
RSP, ELEVP, STAB
SDAC
Pot Power 271125
XMTR Contrl. Column 271027
XMTR Contrl Wheel 271026
XMTR Rudder Pedal 271025
XMTR Elevator Position 271022
XMTR Aileron Rt. Position 271021
XMTR Aileron Lt. Position 271020
XMTR Rudder Position. 271024
573/717
271031
271045
271029
271030
22E11
22F33011
Flight Test Recorder (FTR) & Interface Unit
AP Yaw actuator input ±50mv
or Yaw Damper actuator input ±50mv
GMT Clock (312124, 34205,34206)
Ship's Clock
XMTR Control Wheel Force
XMTR Control Column Force
XMTR Rudder Pedal Rt. Force
XMTR Rudder Pedal Lt. Force
Variable Stop Actuator Position 271040
Rudder Travel Limiter Fault FLC-1 271041
Rudder Travel Limiter Fault FLC-2 271042
ADC Bus – Alt Coarse/Fine, Mach, Airspeed
EFCU Bus – Control Wheel Position
AIRBUS Force Sensors
YAW (Rudd Ped) 271003
CWF (CPT horiz.) 271004
CWF (F/O horiz) 271005
CWF (CPT vert.) 271007
CCF (CPT) 271008

Auto-Pilot Yaw Actuator (APYA) and Yaw-Damper Actuator (YDA) orders Instrumentation block-diagram

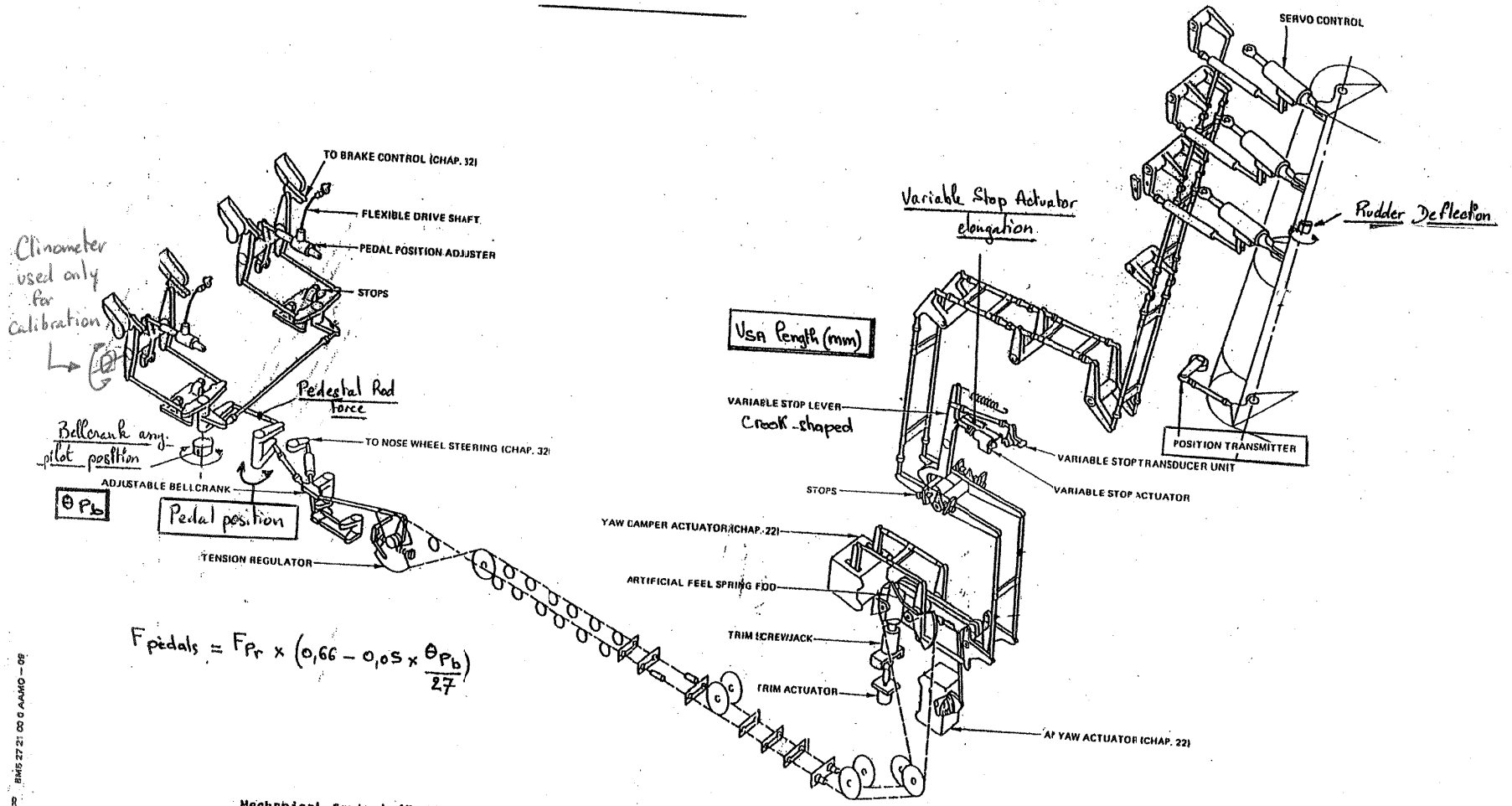


Test configuration and switches position

Test Configuration	CONFIGURATION	Switches position				
		1	2	3	4	5
YDA Controled by A/C System / FAC1	A	/	b	/	b	b
YDA Controled by YDA control-box	B	b	c	/	b	b
APYA Controled by A/C System / FCC1	C	/	/	b	c	c
APYA Controled by APYA control -box	D	c	/	c	c	c

271029 : yaw command (FAC1 output)
271030 : yaw position (FAC1 output)
271045 : fonction generator
271046 : yaw damper actuator input
271047 : AP yaw actuator input
22E11 : ARINC 429 (FAC1 output)
22F33011 : modified input yaw rate

FTI AIRBUS



Mechanical Control (Rudder)
Figure 001

Issue	Date	Approv'd by	Issue	Date	Approv'd by	Effectivity
01	Dec31/81					ST5
02	Dec15/82					
03	Dec30/83					

27-21
P. 3 - 4
Dec 30/83

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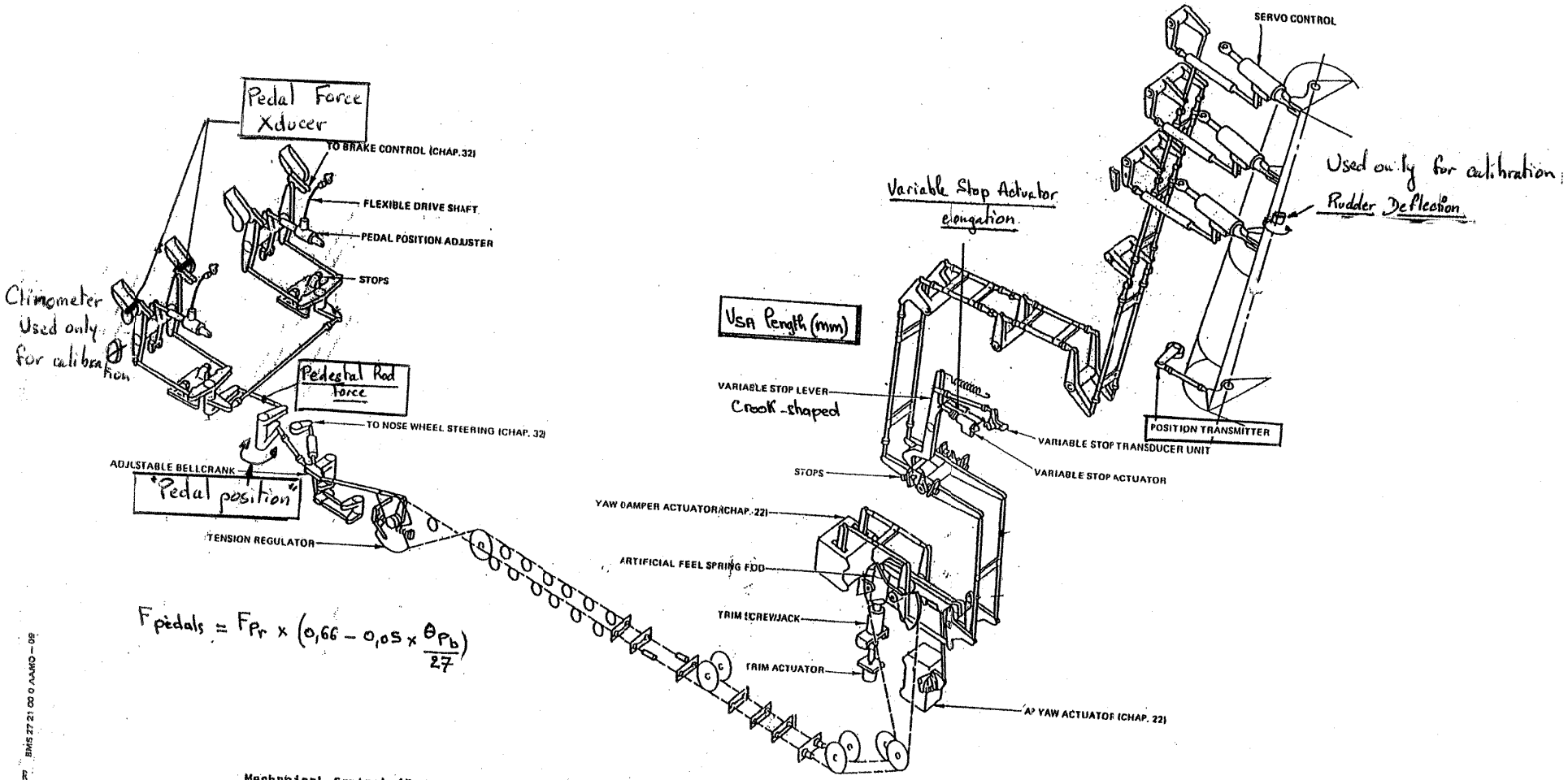
R



$$F_{pedals} = F_{Pr} \times \left(0{,}66 - 0{,}05 \times \frac{\theta_{Pb}}{27}\right)$$

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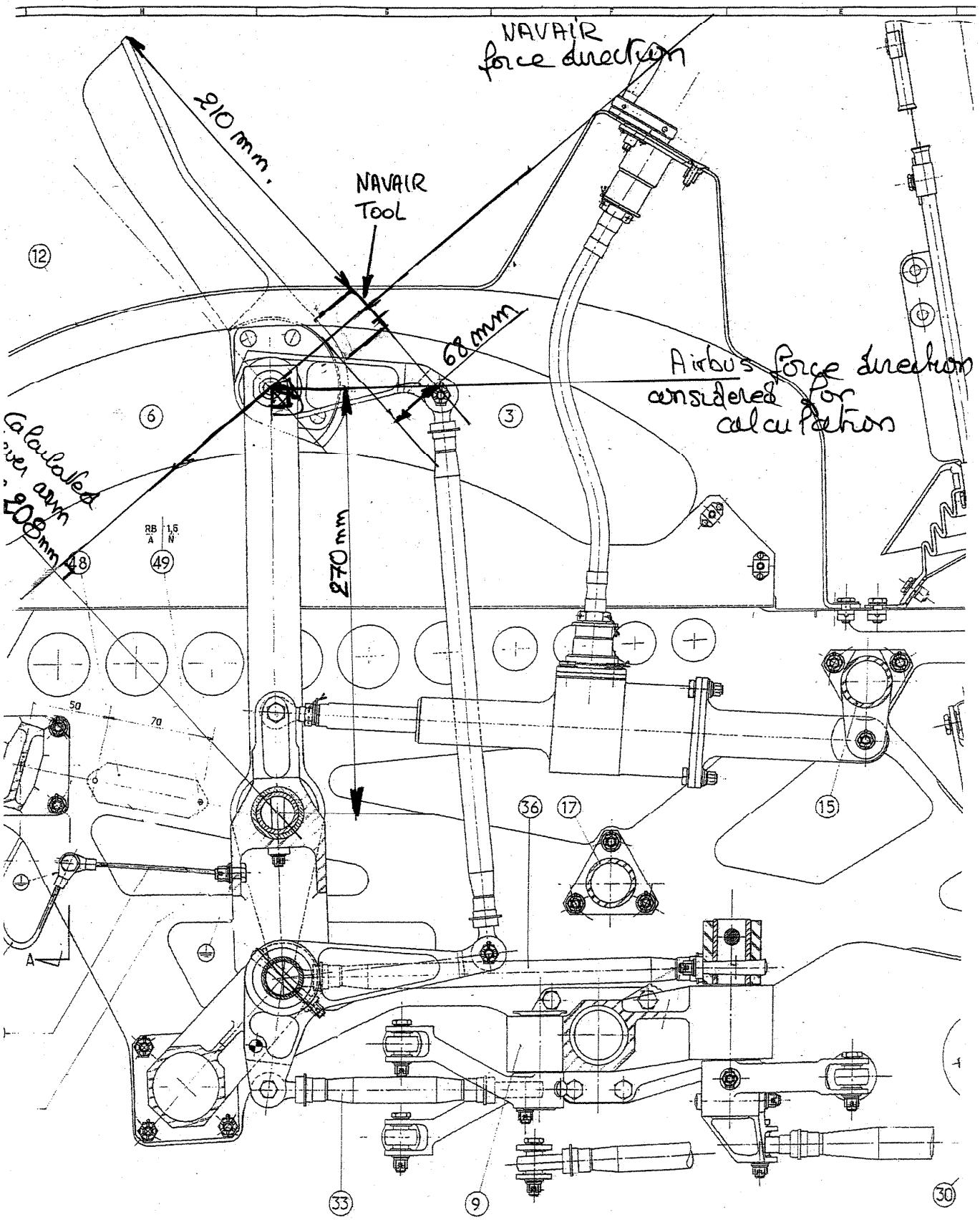
Mechanical Control (Rudder)
Figure 001

Issue	Date	Approv'd by	Issue	Date	Approv'd by	Effectivity
01	Dec31/81					ST5
02	Dec15/82					
03	Dec30/83					


NAVAIR
force direction
210 mm.
NAVAIR
Tool
68 mm
Airbus force direction
considered for
calculation
Calculated
lever arm
209 mm
270 mm
12
6
3
48
49
36
17
15
33
9
30
A270 10320 01 2/2