AIRCRA	A.	RIAL NO.	57	R	EGISTRATION N	0.		OWNER				
\$ 16	76	5036	8		N 56 K			W D				
PILOT		DATE	TYPE		ACFT. HRS.	ENG. ENG. 1	STARTS ENG. 2	TORQUE	LANDINGS	POWER ASSURANCE	ENG. 1	ENG.
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	TOTAL FOR THE DAY				J•					PA.		
	FT & ENGINE IRS / CYCLES / RIN	AIRCR. HOURS			NG. HOURS G. 1 ENG. 2		CYCLES I ENG. 2	TORQUE EVENTS	LANDINGS	I CERTIFY THAT THE DAILY INSPECTION WAS COMPLETE ACCORDING TO THE MAINTENANCE MANUAL.		
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TOTALS I	FOR TODAY:]	5	í	15 115	- 3	3		3			
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A 1965 NO.	INSPECTION									DATE:		
FLIGHT DISCREI	The following maintenan I. Removed all cowling, i Reinstalled all items rei C/W the following insp 12 month eng fire extin month insp. Engines 90 Removed M/R dampers Installed serviceable da Removed power supply Removed power supply Removed T/R paddles F Maintenance. De-fueled aircraft. Serviced nose tire to cor Listalled serviceable wei Removed T/R pitch cha L. Touched up paint in be Removed T/R pitch cha L. Touched up paint in be Repaired: Corrosion on #Iand #2 eng Working rivets on main bea with Sikorsky structural ref Replaced: Leaking T/R gear box outp P/N 76101-05008-101. Hychardware on compressor to Retorques: Stabilizer support fitting bo T/R attach nuts at ACTT: 4:	mspection pan moved after in ections: 12 mg guisher weigh 10 hr/12 mont 12 mg guisher weigh 10 hr/12 mont 12 mg from 12 m	rimed on els, interi- aintenanconth airfra, 12 month airf	this aid or trim this aid or trim trim the cet. arme in the coll of the coll o	ircraft during this in a horizontal stabilizer. sp. 100 hr airframe in control sticks for crace in Avionics insp. 1061-01341, S/N A06 S/N A061-01341, S/N A06 S/N A061-04930, S/N 12 month test/charge. 12 month test/charge. 12 month test/insp. 45-00498 & S/N A24 S/N BA24 S/N A24 S/N BA24 S/N	spection: seats, interio sp, 6 month/1: ks, Engines 30 1-01898, S/N N A061-04360 Installed sam Installed sam 5-00147 for each same as each cannon pl n area, Bulkh 2 engine leaki ft liner, Worki	2 month calenda 10 hr/12 month A061-03495, & . S/N A061-044 e as removed at e after inspectic use of maintena removed. ug. Small punctead at R/H jack ng starter gener ng rivets aft of g bolts at ACT	ar insp items. 12 inspection, Engine S/N A061-0349. 847, S/N A061-0100. Inc. Installed said ture on L/H aft copoint for cracks	month float insp. nes 600 hr/12 9 for overhaul. 11413 TSO: 0.0. me after abin floor panel, in accordance F/R blade boots Corroded			
CORRE	This aircraft and/or compon for work performed. Pertine Signed: For Arthur remains Co. 20 Operational check flight for NAME:	0	performed	ins age	rdance with current F ency under work order bate: 9/30/11 Broussard, red. Satisfact	AA regulation r <u># 11-585</u> LA 70518		d airworthy for r AR527Y		= = 10	pter []	
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ENGINE SERVICE AND

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CALENDAR INSPECTION CHECKLIST	FRE- QUENCY: HOURS/ CAL- ENDAR	CHAPTER/ SECTION/ SUBJECT/ *HELOTRAC	Z O N E	INIT	REMARKS
(m) Radio master system bypass switches (if installed).	12 Months	24-31-00 *231002	1		
28. Gear box chip detector warning system. Test main gear box chip detectors. Main gear box chip detectors and wiring for security and condition.	12 Months	66-64-00 *666403A			Replace any chip detector not meeting specified requirements.
29. Detailed examination of main gear box upper and lower housings for cracks and corrosion. Leakage of oil from area is an indication of a crack.	12 Months	66-10-00 *661002	5		If oil leakage is detected, inspect hous- ings for cracks. Refer to Main Gear Box Hous- ing Inspection, 66-10- 00.
30. Main gear box - main rotor shaft upper bearing shoulder/bearing interface for corrosion. Apply corrosion preventive compound.	12 Months	66-10-00 *661004	5		Refer to Inspection/ Check, 66-10-00, Main Rotor Shaft Corrosion Protection.
NOTE: This inspection is not applicable to gear boxes that have had caulking compound applied at the subject joint during the initial production or at overhaul buildup.					
31. Inspect collective stick position transducer (EEC resolver) electrical connectors per 1500-HOUR INSPECTION, ZONE 5, Item 1.	12 Months	73-21-00 *220001A	5		This inspection is not required if the 1500-Hour collective stick position transducer inspection has been accomplished within the 12 month period.
32. Remove and weigh engine fire extinguishing containers.	12 Months	26-21-00 *262103A/B	4		Refer to Servicing, 26-21-00.
33. Portable fire extinguisher, weight.	12 Months	26-22-00 *262201A/B	1		Minimum weight on extinguisher data plate.
34. Intermediate gear box for corrosion and service related damage.	12 Months	66-20-00 *662004	3		For inspection procedure and accept/reject criteria, refer to Intermediate Gear Box Corrosion Inspection, 66-20-00.

5-20-00

Page 74 Oct 31/11 1500-Hour Inspection Checklist Zone 5.

	CILARTER /			
1500-HOUR INSPECTION CHECKLIST ZONE 5	CHAPTER/ SECTION/ SUBJECT/ *HELOTRAC	ZONE	INIT	REMARKS
	*051505			All 1500 Hour Zone 5 Code
1. Cyclic and collective stick transducers for security and condition. Electrical wiring and connectors for security and condition. Disconnect collective stick position transducer (EEC resolver) electrical connectors P1 and P2 from their respective mating connectors J817 and J818.	22-10-08 22-20-02 73-21-00 *220001	5		Inspect every 1500 hours or 12 months, whichever occurs first. Also, refer to Honeywell Manual for helicopters with DAFCS).
(a) Visually inspect connectors P1 and P2 for damaged or missing pins or sealing plugs. All cavities should be filled with pins or sealing plugs.				Missing pins or sealing plugs should be readily visible but a magnifying glass may be helpful for FOD or water intrusion.
(b) Inspect connectors P1, P2, J817, and J818 for FOD (dust, crystalline objects, metal particles) and water intrusion or corrosion.				
(c) Thoroughly clean and remove any FOD. Remove any water droplets and corrosion. (Refer to Corrosion Control Manual, SA 4047-76-8.)				
(d) Replace damaged or missing pins/sealing plugs. (Refer to Airframe Wiring Harnesses, 20-31-00.)				Empty cavities are filled with sealing plugs, MS27488-20, or pins, MS39029/58-363.
2. Pitch, roll, and collective gradient spring cylinders or links, and magnetic brakes or trim servos for security and general condition. Electrical connectors for security. Check selfaligning bearings in pitch, roll, and collective gradient spring cylinders or links. Remove attachment hardware at end opposite magnetic brakes or trim servos, check self-aligning bearings at both ends for freedom. Be sure antirotation washers are reinstalled and install bolts, washers, nuts, and cotter pins.	67-13-01 67-13-02 67-13-03 *671301	5		Refer to Honeywell Manual (helicopters equipped with DAFCS).

5-20-00