ATTACHMENT 6

AIRWORTHINESS GROUP CHAIRMAN'S FACTUAL REPORT

CEN13FA192

Bell Helicopter Broussard Work Order S76T-137-00708 (71 Pages)

Traveler Cover Sheet

PARTS ISSUED
INITIAL

Date Received: 2/10/2012 **Customer:** ERA HELICOPTERS

Part#: 76101-05101-041 **Serial #** A137-00708X

Main Rotor	Tail Rotor	Tip Cap	Main Yoke
	X		

Work Order# S76T-137-00708 **AOG** _____ **Reg** X

- [x] All Items Signed Off
- [ ] Parts Verified
- [x] Final Inspections Completed
- [ ] 8130 Completed

Overhaul Manual/Repair Manual Must Be Readily Available Utilized, And Is The Control Document To Accomplish Blade Repair

Inspection Required Prior to Paint Prep

_____

DUE

- [ ] NEW BOX REQUIRED
- [ ] NEW CONTOURS

Bell Helicopter Textron Inc.
Work Order Traveller
WORK ORDER ID: S76T-137-00708/1

Work Order ID	Part ID	Release	WantDate	Start Date	Finish Date	Qty	Drawing ID/Rev Engineer
S76T-137-00708/1	TRLAMDRP	1/13/2012	1/13/2012			1.00	

TAIL ROTOR LA 212, 412 BLADE

Order ID:
Part ID:

Customer:

Ship To:

Shop Resources Type	Seq#	End Qty	Start Qty	Setup Hrs	Run Hrs	Start Date	Finish Date	Drawing ID/Rev
LA BLADE REPAI	10	1.00	1.00	0.00	0.00			

Bell Helicopter Textron Inc.
Tail Rotor Blade
76101-05101-041
DISCREPANCY/QUOTE SHEET

Customer: Era Aviation

Part No: 76101-05101-041 Serial No: A137-00708x

Blade T.T: 14,800:5 Date Inspected: 1-16-2012 Inspector: [redacted]

	Repair Data: Inspected and repaired per Sikorsky Specifications Inspected I.A.W. CMM SA 4047-76-5 Table 2-2,	*Part Number*	Repair Procedure # Composite Materials	Repair Procedure # O R I	Verified Complete	Completed By
Discrepancy	Polyurethane supplemental strips eroded on A & B paddles	76101-05011-102	CMM SA-4047-76-5 proc 10 (4-16).		[redacted]	[redacted]
Corrective Action	Removed and Replaced polyurethane supplemental strips on A & B paddles	Qty 2				
Discrepancy	Rubber boot torn on A & B paddles	76101-05008-101 Qty 2	MM SA 4047-76-2 (65-21-01) page 203			
Corrective Action	Installed boot on A & B paddles					
Discrepancy	Comply with 500 hr inspection		ASB 76-65-60			
Corrective Action	Complied with (spar inspection 500 hour)					
Discrepancy	Comply with 1500 hr inspection		SA-4047-76-2-1 checklist zone 3, item 4-4A			
Corrective Action	Complied with 1500 hr inspection					
Discrepancy						
Corrective Action						
Discrepancy						
Corrective Action						
Discrepancy						
Corrective Action						
Discrepancy						
Corrective Action						

Continued on Next Page

Bell Helicopter Textron Inc.
Tail Rotor Blade
76101-05101-041
DISCREPANCY/QUOTE SHEET

Customer: Error! Not a valid link. Era Aviation Serial No: A137-00708x Error! Not a valid link.

Repair Data: Inspected and repaired per Sikorsky Specifications		Part Number	Repair Procedure # Composite Materials	Repair Procedure # ORI	Verified Complete	Completed By
Discrepancy						
Corrective Action						
Discrepancy / Corrective Action	☐ Perform RII inspection (when required) ☐ Accomplished all applicable PHI RII Requirements					
Discrepancy	☑ Refinish per CMM SA4047-76-5 Procedure # 6 4-10					
Corrective Action	☑ Refinished per CMM SA4047-76-5 Procedure # 6 4-10					
Discrepancy	☑ Balance					
Corrective Action	☑ Balanced Span A Ø Span B Ø Chord A 14.5 Chord B 65.1					

[redacted]

TRACK	MINUTES			
BLADE SERIAL NO.	A137-00708x			
BLADE DWG. NO.	76101-05101-041			
MFG. 5-16-84	RS	-040C-I	RS	
REP.	RS	-018M-I	RS	
REP.	RS	-022L-IV	RS	
REP.	RS		RS	
REP.	RS		RS	

Bell Helicopter Textron Inc.

Repair Station # R8DR713Y

CUSTOMER BLADE RECEIVER

Customer Order # _____

Work Order # _____

DO NOT MIX CUSTOMER SHIPMENTS USE ONE RECEIVER FOR EACH CUSTOMER AND BLA

Customer: FLA AVIATION Date Received: JAN 1 2 2012 Received By: [redacted]

TYPE			Part Number	Serial Number	Time Since New	Records Yes	Records No	Date Shipped	Box Number
M/R	T/R	T/C							
	X		7612F05101-041	A137-00705	14522.5	X			[illegible] DX

LOCATION OF CONTAINERS:

Condition of Container:	☐ **Good** ☐ **Fair** ☐ **Repairable**	☐ **Blade Box Needs To Be Replaced**
Condition of Contours:	☐ **Good**	☐ **Replace Contours**

To insure that Containers are returned to the proper Customer be sure to accurately mark each container with Customer Name and Box # listed on this form.

Work to be performed: STANDARD REPAIR

Comments/Special Instructions: _____

Incoming Paint Scheme: Select One _____ High Visibility Paint Scheme _____ Standard Paint Scheme

Freight in: UPS Freight out: _____ Contact Name: _____

Purchase order Information: _____ Telephone: _____ Fax: _____

When completed forward a copy to the General Manager

Customer support must be notified if the container is not suitable for shipment of repaired items. The Original completed CUSTOMER BLADE RECEIVER, and all documentation removed from the container is put into a folder and forwarded to the Quality Control Department for incoming blade inspection.

RBI –L – 001

COMPONENT MAINTENENCE RECORD

DATE	REPAIRS OR OVERHAULS	SIGNATURE	CERTIFICATE NO.
	10-25-07 W/O 36671 X-rayed per MIL-STD-453, ultrasonic inspected per QATI 3098, 3261 & 3291, performed all scheduled inspections up to and including 1500 hr. inspection in accordance with composite MM 4047-76-5 table 3-2 and MM 4047-76-2, 65-21-00 and 65-21-01, complied with AD 94-14-20 R-1 & 89-07-12, ASB 76-65-40A, 76-65-41, 76-65-43B, 76-65-46 & 76-65-60; Performed 4000 hour torque check per RS-022N-III; Replaced both pitch horns per RS-040C-I; Replaced both primary poly strips per RS-018M-1; Replaced both secondary poly strips per SA 4047-76-5 section 4-16; Replaced laminar pad on spar of paddle B on outboard & pylon side per RS-022L-IV; Repaired skin damage on paddle B pylon side at sta. 9.0 to sta. 10.25 per OREI 2007-29; Replaced and reassembled center hardware, 2 ea fairings, 4 ea pivot bearings & 4 ea spandwise brackets per MM SA 4047-76-2 section 65-21-01; Refinished paint per drwg. 76101-05020; Static balanced per drwg. 76101-11528 amdt. 20 and drwg. 76101-11529 @ T.T. 12222.3	[redacted]	[redacted]

...AFT

ASSOCIATE LIFE LIMITED EQUIPMENT TO MAJOR COMPONENT

1. MAJOR COMPONENT	2. PART NUMBER	3. SERIAL NUMBER
TAIL ROTOR BLADE	76101-05101-041	A137-00708

65-1250

PART NAME A.	PART NUMBER B.	SERIAL NUMBER C.	DATE D.	PERFORM. ORGAN-IZATION E.	TIME SINCE NEW ON ITEM SHOWN IN BLOCK 1	TIME SINCE NEW ON LIFE LIMITED ITEMS SHOWN IN COLUMN A. INSTALLED	REMOVED	REASON FOR REMOVAL
Spar Fastener	76101-05023-105	NONE	6-22-84	Sikorsky	NEW	NEW		
Spar Fastener	76101-05023-107	NONE	"	"	"	"		
Spar Fastener	76101-05023-108	NONE	"	"	"	"		
Spar Fastener	76101-05023-109	NONE	"	"	"	"		
Spar	76101-05017-045	A116-01207	"	"	"	"		
Blade	76101-05020-045	1819	"	"	"	"		
Blade	76101-05020-045	1820	"	"	"	"		
Pitch Horn	76101-05006-042	1819	"	"	"	"	12222.3	time X-pired
Pitch Horn	76101-05006-042	1820	"	"	"	"	12222.3	time X-pired
Pitch Horn	76101-05006-042	B361-00269	10-25-07	CTI	12222.3	New		
Pitch Horn	76101-05006-042	B361-~~00269~~ 00328	10-25-07	CTI	12222.3	New		

1819

1820

B361-00269

B361-00328

UNITED TECHNOLOGIES SIKORSKY AIRCRAFT

AIRCRAFT MAINTENANCE LOG

T/R BLADE P/N 76101-05101-041 S/N A137-00708

REMARKS - PILOT OR INSPECTOR	MAINTENANCE WORK	SIGNATURE	DATE
11-7-91 C/W AD 84-06-02, ASB 76-65-35A INSPECT OF - INSPECTED EDGE OF SPAR I/A/W SIKORSKY MSG # CBT-TC-91-1512 AND NO CRACKS FOUND BLADE T.T. 4364:48 A/C T.T. 4364:48	SPAR PLUG - FOUND	[redacted]	11-7-91
3-27-92 4497:27 A/C TT	C/W AD 84-06-02 INSP T/R BLADE ELLIPTICAL SPAR PLUG.	[redacted]	3-27-92
11-20-92 4788:20	C/W AD 84-06-02 INSP. OF T/R BLADE ELLIPTICAL SPAR PLUG	[redacted]	11-20-9
7-25-93 TT: 5078:45	C/W ASB 76-65-43 INSP. Bumper pads For proper gaps Red - I.B .107 Yellow I.B .105 O.B .100 O.B .93	[redacted]	7-25-9
10-12-93 TT 5229:22	C/W ASB 76-65-44 T/R BLADE FAIRING MODIFICATION	[redacted]	10-12-93
	C/W AD 84-06-02 ELLIPTICAL PLUG INSPECTION	[redacted]	10-12-93
6-2-94 TT: 5721:05	C/W AD 84-06-02 INSPECTION OF Tail Rotor Blade Elliptical SPAR Plug.	[redacted]	6-2-9

SIKORSKY AIRCRAFT Division of UNITED TECHNOLOGIES

AIRCRAFT MAINTENANCE LOG

REMARKS - PILOT OR INSPECTOR	MAINTENANCE WORK	SIGNATURE	DATE
8-27-94	I/A/W AD 94-14-20 BY INSTALLATION OF 76102-05004-111 RETAINING PAD I/A/W 76-65-35A PARA 2C. AD 84-06-02 NOW doesn't APPLY	[redacted]	8-27-94
8-30-96	C/W insp of T/R blades I/A/W Zone 3, Item 7 - Repainted T/R blade assy at comp IT 7028:31	[redacted]	[redacted]
11-19-98 total time (REF-T 7712.5)	C/W AD 94-14-20 T/R [redacted] plug insp.	[redacted]	11-19-98
8-10-01 X-rayed, ultrasonic inspected per QATI 3016 and 3098, performed 1500 hour inspections, inspected pitch horns per AD 89-07-12, complied with ASB 76-65-41A, 76-65-43 & 76-65-46; Replaced both supplemental polyurethane strips per SA 4047-76-5 section 4-16 rev. 9; Replaced both primary polyurethane strips per RS-018M-1; Replaced pivot retainers 4 ea per RS-22L-V; Replaced pivot bearings 4 ea per MM SA 4047-76-2 section 65-21-01; Repaired skin on paddle A per RS-22M-VII; Repaired wire mesh and performed 4000 hr torque check per RS-018-III and RS-002N-III; Replaced tips on both paddles per RS-021C-II; Replaced and reassembled center hardware per MM SA 4047-76-2 section 65-21-01; Refinished paint per Drwg. No. 76101-05020; Static balanced per drwg. 76101-11528 admt. 20 and drwg. 76101-11529 rev. C @ TTSN 7905.1 Pertinent details pertaining to the inspection, alteration and/or repairs are on file at this repair station under W.O. 29506 FAA CRS RW1R467K.		[redacted]	[redacted]	8-10-01
10/11/01 575EH ACTT 1082.3 -	C/W AD 94-14-20 & ASB 76-65-35B by Arrow Aviation - WO - 01-177	[redacted]	12-30-03
12-30-03 X-rayed, ultrasonic inspected per QATI 3016 and 3098, performed all scheduled inspections up to and including 1500 hr. Inspection in accordance with Composite Materials Manual 4047-76-5 table 2-2 and Maintenance Manual 4047-76-2, 65-21-00 and 65-21-01, inspected elliptical plug for debonding per A.D. 94-14-20, complied with ASB 76-65-43 & 76-65-46; Replaced spar plug per RS-022L-II; Repaired trailing edge damage on paddle A per RS-018M-IV; Repaired gouge in pitch horn bushing area on both paddles per ORI 76100-018 part VI; Replaced both primary polyurethane strips per RS-018M-1; Replaced both supplemental polyurethane strips per SA 4047-76-5 section 4-16 rev. 9; Replaced and reassembled center hardware per MM SA 4047-76-2 section 65-21-01; Refinished paint per Drwg. No. 76101-05020; Static balanced per drwg. 76101-11528 admt. 20 and drwg. 76101-11529 rev. C @ TTSN 9663.9 All repairs were performed utilizing designed, process specification, drawings, manuals or other media supplied by the manufacture. Pertinent details to the inspection, alteration and repairs are on file at this repair station under W.O. 32570 FAA CRS RW1R467K.		[redacted]	[redacted]

1. Approving National Aviation Authority/Country:	2.	3. Form Tracking Number:
UNITED STATES	**AUTHORIZED RELEASE CERTIFICATE** FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	36671

4. Organization Name and Address:	5. Work Order/Contract/Invoice Number:
Composite Technology INC. 1001 Ave. R, Grand Prairie, TX 75050 FAA Certified Repair Station Number RW1R467K	RO245-000982-2007

6. Item:	7. Description:	8. Part Number:	9. Eligibility:*	10. Quantity:	11. Serial/Batch Number:	12. Status/Work:
1	Tail Rotor Blade	76101-05101-041	N/A	1	A137-00708X	REPAIRED

13. Remarks: X-rayed per MIL-STD-453, ultrasonic inspected per QATI 3098, 3261 & 3291, performed all scheduled inspections up to and including 1500 hr. inspection in accordance with composite MM 4047-76-5 table 3-2 and MM 4047-76-2, 65-21-00 and 65-21-01, complied with AD 94-14-20 R-1 & 89-07-12, ASB 76-65-40A, 76-65-41, 76-65-43B, 76-65-46 & 76-65-60; Performed 4000 hour torque check per RS-022N-III; Replaced both pitch horns per RS-040C-I; Replaced both primary poly strips per RS-018M-1; Replaced both secondary poly strips per SA 4047-76-5 section 4-16; Replaced laminar pad on spar of paddle B on outboard & pylon side per RS-022L-IV; Repaired skin damage on paddle B pylon side at sta. 9.0 to sta. 10.25 per OREI 2007-29; Replaced and reassembled center hardware, 2 ea fairings, 4 ea pivot bearings & 4 ea spandwise brackets per MM SA 4047-76-2 section 65-21-01; Refinished paint per drwg. 76101-05020; Static balanced per drwg. 76101-11528 amdt. 20 and drwg. 76101-11529 @ T.T. 12222.3
Full Details on W.O. 36671.

14. Certifies the items identified above were manufactured in conformity to:	19. ☒ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 13
☐ Approved design data and are in condition for safe operation. ☐ Non-approved design data specified in Block 13.	Certifies that unless otherwise specified in block 13, the work identified in Block 12 and described in Block 13 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

15. Authorized Signature:	16. Approval/Authorization No.:	20. Authorized Signature: [redacted]	21. Approval/Certificate No.: [redacted]
17. Name (Typed or Printed):	18. Date:	22. Name (Typed or Printed): Linda J. Gray	23. Date (m/d/y): 10-25-07

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the part/component/assembly.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness accepts parts/components/assemblies from the airworthiness authority of the country specified in Block1.

Statements in Blocks 14 and 19 do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification [illegible] national regulations by the user/installer before the aircraft may be flown.

ERA HELICOPTER [redacted]

FAA Form 8130-3 (6-01) *Installer must cross-check eligibility with applicable technical data.

U4-10-10

SIKORSKY AIRCRAFT Division of UNITED TECHNOLOGIES

COMPONEN[illegible]G CARD

PART NO. 76101-05101-041 NOMENCLATURE TAIL ROTOR BLADE SERIAL NO. A137-00708X

651700

AIRCRAFT NO.	DATE INSTALLED	DATE REMOVED	AIRCRAFT TIME	COMPONENT TIME T.S.O.	COMPONENT TIME T.T.	REASON FOR REMOVAL
760274	30Nov.1984		NEW	NEW	NEW	
760274		2 OCT 87	2230.0	New	2230.0	C/W MG-65-40
760274	13 Nov 87		2315.9	New	2230.0	ERROR
760274		17NOV 90	3915.0	NEW	3915.0	REPAINT
760274	29NOV 90		3915.0		3915.0	[redacted]
760274		6-22-01	3991.0	NEW	3991.0	INSPECT & REPAIR

SA 7343-9 REV A.

OVERHAUL INTERVAL

RETIREMENT TIME 12,000 HRS

Era HELICOPTERS LLC
a SEACOR company

Repair Order

Mail Invoices To :

ERA HELICOPTERS LLC
600 Airport Service Road
LA Lake Charles
USA 70605

Repair Order # : RO245-011863-2012
Amendment # : 0
THIS NUMBER MUST APPEAR ON ALL INVOICES, PACKING SLIPS, PACKAGES & CORRESPONDENCE

Supplier : ROTOR BLADES INC
580 ST ETIENNE RD, STE H
BROUSSARD, LA 70518 US

ALEXIS ROBIN
Phone # : [redacted]
Fax # : [redacted]

Ship To : ERA HELICOPTERS LLC
600 AIRPORT SERVICE ROAD
LAKE CHARLES, LA 70605 USA

Supplier #	RO Date	Ship By	F.O.B	Pay Term
8100-8100-S3	01/09/2012	UPS GRND	ORIGIN	NET 30
Contact		**Phone #** **Ref Doc #**	**Quotation #**	**Approved By**

Line #	Part # / Description	Serial # / Lot #	Due Date	Qty.	UOM	Repair Cost
1	76101-05101-041 TAIL ROTOR BLADE /	A137-00708X	02/09/2012	1	EA	

Repair Order # : RO245-011863-2012

Maintenance Information Details

Part # : 76101-05101-041 **Part Desc :** TAIL ROTOR BLADE **Serial # :** A137-00708X

Component Details

Component ID	ROT-853	Level Code		TSN	14800.5	TSI	312.1
EIPN Comp ID		Position Code		TSO	14793.3	TSR	14793.3

Parameter Details (FH)

Task Details

Work unit # / Work Unit Desc	Maintenance Type / Ref Document #	LPD / LPV	Current Value	Calender Interval / Usage Interval	Remaining Value
AD86-19-14	Others			32263 Days	31769 Days
Replacement of listed serial numbered tail rotor assemblies					
AD94-14-20 R1	Others			32263 Days	31769 Days
Inspection of the tail rotor blade spar elliptical centering plug for disbonding and the addition of a retaining pad on the pitch change shaft					
S76 TR BLADE INSP	Inspection	07/06/2011	14800.5 HR		187.9
Tail Rotor Blade Boroscope Inspection		14488.4 HR		500 HR	
S76 TRB SPAR PLUG INSP	Inspection		14800.5 HR		17.4
Tail Rotor Blade Spar Plug Inspection		13317.9 HR		1500 HR	

Part # : 76101-05020-045 **Part Desc :** TAIL ROTOR BLADE **Serial # :** 1819

Component Details

Component ID	ROT-4121	Level Code	1.1	TSN	5136.6	TSI	5129.4
EIPN Comp ID		Position Code	T/R BLADE 1	TSO	5129.4	TSR	5129.4

Parameter Details (FH)

Task Details

Work unit # / Work Unit Desc	Maintenance Type / Ref Document #	LPD / LPV	Current Value	Calender Interval / Usage Interval	Remaining Value
S76 TR BLADE RET	Retire		5136.6 HR		14863.4
Tail Rotor Blade Retirement				20000 HR	

Part # : 76101-05020-045	Part Desc : TAIL ROTOR BLADE	Serial # : 1820

Component Details

Component ID	ROT-4122	Level Code	1.2
EIPN Comp ID		Position Code	T/R BLADE 2

Parameter Details (FH)

TSN	5136.6	TSI	5129.4
TSO	5129.4	TSR	5129.4

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
S76 TR BLADE RET	Retire		5136.6 HR		14863.4
Tail Rotor Blade Retirement				20000 HR	

Part # : 76101-05006-042	Part Desc : HORN ASSY, TR	Serial # : B361-00269

Component Details

Component ID	ROT-4123	Level Code	1.3
EIPN Comp ID		Position Code	T/R HORN 1

Parameter Details (FH)

TSN	2578.2	TSI	2578.2
TSO	2578.2	TSR	2578.2

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
AD89-07-12.	Others		2578.2 HR		9421.8
Establish a life limit for tail rotor horn, P/N 76101-05006 of 12,000 hours time in service, remove all items that have exceeded this value.				12000 HR	
S76 TR HORN RET	Retire		2578.2 HR		9421.8
Tail Rotor Horn Retirement				12000 HR	

Part # : 76101-05006-042	Part Desc : HORN ASSY, TR	Serial # : B361-00328

Component Details

Parameter Details (FH)

Component ID	ROT-4124	Level Code	1.4	TSN	2578.2	TSI	2578.2
EIPN Comp ID		Position Code	T/R HORN 2	TSO	2578.2	TSR	2578.2

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
AD89-07-12.	Others		2578.2 HR		9421.8
Establish a life limit for tail rotor horn, P/N 76101-05006 of 12,000 hours time in service, remove all items that have exceeded this value				12000 HR	
S76 TR HORN RET	Retire		2578.2 HR		9421.8
Tail Rotor Horn Retirement				12000 HR	

8130-3 REQUIRED

Please Inspect/Repair/Overhaul as required in accordance with Manufacturer's Manual and current Regulatory Requirements.

N578EH 1500 hr isnpection due paint erroded inspect and repair

Please email quote with turn around time to roquotes@erahelicopters.com

Legends : **F.O.B- Free On Board, QTY-Quantity, UOM-Unit Of Measurement, EIPN-End Item Part Number, TSN- Time Since New, TSI-Time Since Inspection, TSO-Time Since Overhaul, TSR-Time Since Repair, LPD-Last Performed Date, LPV-Last Performed Value.**

THIS ORDER IS EXPRESSLY MADE AND ITS ACCEPTANCE IS EXPRESSLY LIMITED TO THE TERMS AND CONDITIONS CONTAINED HEREIN AND ATTACHED TO LAST PAGE		TOTAL :
	AUTHORIZED SIGNATURE	

Era HELICOPTERS LLC a SEACOR company

600 Airport Service Road
Lake Charles LA 70605

Form M-003

REPAIRABLE

Part Description:		Part Number:	Serial Number:	Comp. ID:
TAIL ROTOR BLADE		76101-05101-041	A137-00708X	ROT-853
TSN: 14800.5	TSO: 14793.3	Next Maintenance: S76 TRB SPAR PLUG INSP	Next Maintenance Due: 14818FH	
Object Type: Tracked Component	Ref. Doc. No: TL-042657-2012	Date: 01/07/2012	PO Number: RO245-000982-2007	

****REMOVED REPAIRABLE DETAILS****

A/C Registration Number:	NHA Part Number:	NHA Serial Number:	Position Code:
N578EH			T/R BLADE ASSEMBLY 1
Component Replacement No: TLCR-010700-2012	Removed By: Hudson, David W.	Removal Station: LAKE CHARLES	Removal Date: 01/07/2012
Removal Code: EROSION	Reason for Removal Remarks: 1500 hr inspection due, paint erroded		
Comments:			

[X] Era Owned Part [ ] Vendor Owned Part [ ] Warranty Core Warranty Claim No:

Date Generated: 01/07/2012

Era HELICOPTERS LLC a SEACOR company

600 Airport Service Road
Lake Charles LA 70605

Form M-003

REPAIRABLE

Part Description:		Part Number:	Serial Number:	Comp. ID:
TAIL ROTOR BLADE		76101-05101-041	A137-00708X	ROT-853
TSN: 14800.5	TSO: 14793.3	Next Maintenance: S76 TRB SPAR PLUG INSP	Next Maintenance Due: 14818FH	
Object Type : Tracked Component	Ref. Doc. No : TL-042657-2012	Date: 01/07/2012	PO Number: RO245-000982-2007	

****REMOVED REPAIRABLE DETAILS****

A/C Registration Number:	NHA Part Number :	NHA Serial Number:	Position Code:
N578EH			T/R BLADE ASSEMBLY 1
Component Replacement No: TLCR-010700-2012	Removed By: Hudson, David W.	Removal Station: LAKE CHARLES	Removal Date: 01/07/2012
Removal Code: EROSION	Reason for Removal Remarks: 1500 hr inspection due, paint erroded		
Comments:			

[X] Era Owned Part [ ] Vendor Owned Part [ ] Warranty Core Warranty Claim No:

Date Generated: 01/07/2012

SIKORSKY AIRCRAFT Division of UNITED TECHNOLOGIES — ERA HELICOPTERS — 884750 — AUG 13 2001 — COMPONENT A137-00708X

PART NO. 76101-05101-041 NOMENCLATURE TAIL ROTOR BLADE SERIAL NO. A137-00708X

AIRCRAFT NO.	DATE INSTALLED	DATE REMOVED	AIRCRAFT TIME	COMPONENT TIME T.S.O.	COMPONENT TIME T.T.	REASON FOR REMOVAL
760274	30Nov.1984		NEW	NEW	NEW	
760274		2 OCT 87	2230.0	New	2230.0	c/w 76-65-40
760274	13 Nov 87		2315.9	New	2230.0	
760274		17NOV 90	3915.0	NEW	3915.0	REPAINT
~~760274~~	~~29NOV 90~~		3915.0		3915.0	
760274		11-17-90	3915.0		3829.1	repaint
N579EH 760274	11-29-90		3915.0		3829.1	
N579EH 760274		6-22-01	7991.0		7905.1	erosion
575EH	10-9-01		1082.3		7905.1	
575EH	—	10-2-03	2541.1	~~9603.9~~ ~~1700.8~~	9603.9	Erosion
N578EH	6-16-04	—	5669.4	—	9663.9	
578EH	—	2/10/07	8227.8	—	12222.3	Due Removal
578EH	11-24-07	—	9327.3		12222.3	
578EH	—	1-7-2012	11905.5	tsi-312.1	14800.5	paint erosion; 1500hr plug/insp

[redacted]
ERA HELICOPTERS
JAN 05 2004
RECEIVING INSP.

SA 7343-9 REV A.

64-10-10

OVERHAUL INTERVAL

RETIREMENT TIME

SIKORSKY AIRCRAFT — UNITED TECHNOLOGIES

COMPONENT MAINTENANCE RECORD

DATE	REPAIRS OR OVERHAULS	SIGNATURE	CERTIFICATE NO.
	Mfg. Date of Tail Rotor Blade #1: 6-14-84		
	#2: 6-14-84		
	Mfg. Date of Spar: 5-16-84		
	A.D. 84-06-02, Amd. 39-4829 - Inspection of Elliptical Centering Plug and Installation of		
1September1984	Retaining Pad (A.S.B. 76-65-35A)	[redacted]	[redacted]
~~3 FEB87~~ ~~31 MAR 84~~	AD 86-19-14 ASB 76-65-38 INSPECTION OF T/R BLADE; RE-	N/A by S/N	
	Moval of blade from SERVICE by Ser. Number	[redacted]	[redacted]
SEPT 15, 1986	ASB 76-65-37 REINSPECTION OF T/R ~~BLADE~~ BLADE ASSY		
	N/A by ~~Serial Number~~ COMPONENT TIME	[redacted]	[redacted]
OCT 28 1986	ASB 76-65-39 INSPECT T/R bladespar for dis-	[redacted]	[redacted]
	bonding.		
[illegible]	BLADE SPAR ASSY WAS INSPECTED PER S/A		
	ASB 76-65-40 AND FOUND TO MEET THE		
	MANUFACTURE'S INSPECTION CRITERIA	[redacted]	[redacted]
4-25-88	C/W AD 84-06-02 / ASB 76-65-35A INSP?		
	SPAR ELIPTICAL PLUG		
10/24/88	C/W AD 84-06-02 AMND SPAR ELIPTICAL PLUG		
12/29/88	ASB 76-65-35A Tail Blade ELLIPTICAL Plug Insp.		
11/17/90	SANDED + REPAINTED BLADES I/A/W SIKORSKY INSTRUCTIONS		
11/17/90	C/W 500HR INSP I/A/W COMPOSIT MANUAL INCLUDING		
11/17/90	AD 84-06-02 T/R SPAR PLUG INSP C/W		

TAIL ROTOR BLADE - MAINTENANCE PRACTICES

1. Removal/Installation Tail Rotor Blades.

   Refer to Tail Rotor - Removal/Installation, 65-21-00.

2. Inspection/Check Tail Rotor Blades.

   A. General Inspection Requirements. For detailed inspection requirements, accept reject criteria for damage, and repair limits for tail rotor blades, refer to S-76 COMPOSITE MATERIALS MANUAL, SA 4047-76-5.

   B. Overspeed Inspection Requirements. For a definition of overspeed applicable to the tail rotor blades and for inspection requirements that must be complied with after overspeed, refer to Inspection/Check 65-12-00.

   C. Sudden Stoppage Inspection Requirements. For a definition of sudden stoppage applicable to the tail rotor blades and for inspection requirements that must be complied with after sudden stoppage, refer to Inspection/Check, 65-12-00.

   D. Elastomeric Bearing (Tail Rotor Snubber) Inspection Procedures.

      NOTE: The term "eraser-type" is used to describe the wear of elastomeric bearings because of the resemblance to the result of using a soft eraser. This pile-up of shredded rubber is typical of the wear of the elastomeric bearings (Figure 201).

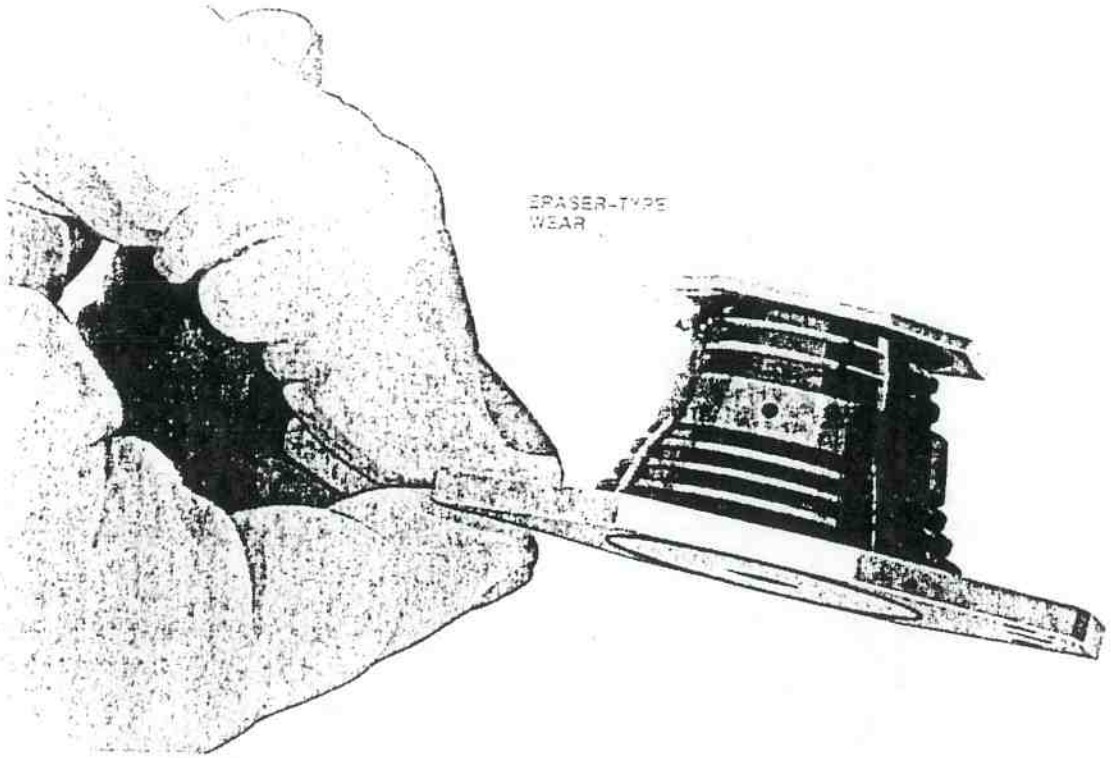
      (1) At 1500-hour inspection interval, remove pivot bearings and do a visual inspection. The principal area susceptible to elastomer wear is in the elastomer layer closest to the blade spar. Replace snubbers showing eraser-type wear (Figure 201). Also inspect for separation in the bearing by deflecting the bearing as shown in Figure 202. Replace snubber if any separation is detected.

3. Cleaning/Painting/Ice Removal Tail Rotor Blades.

   A. Cleaning.

      (1) Procedures and materials for cleaning tail rotor blades are the same as for main rotor blades. (Refer to Cleaning/Painting/Ice Removal, 65-11-00.)

      CAUTION: TO PREVENT DAMAGE TO ELASTOMERIC BEARINGS, DO NOT ALLOW SOLVENTS, GREASE, LUBRICATING OILS, OR SILICON SPRAYS TO CONTACT THEM.



S76A_55450 (B)
SA

Snubber - Example of Eraser-Type Wear (None Acceptable)
Figure 201

(2) Procedures and materials for cleaning tail rotor blade elastomeric pivot bearings are the same as for main rotor elastomeric bearings. (Refer to Cleaning Elastomeric Bearings, 65-12-05.)

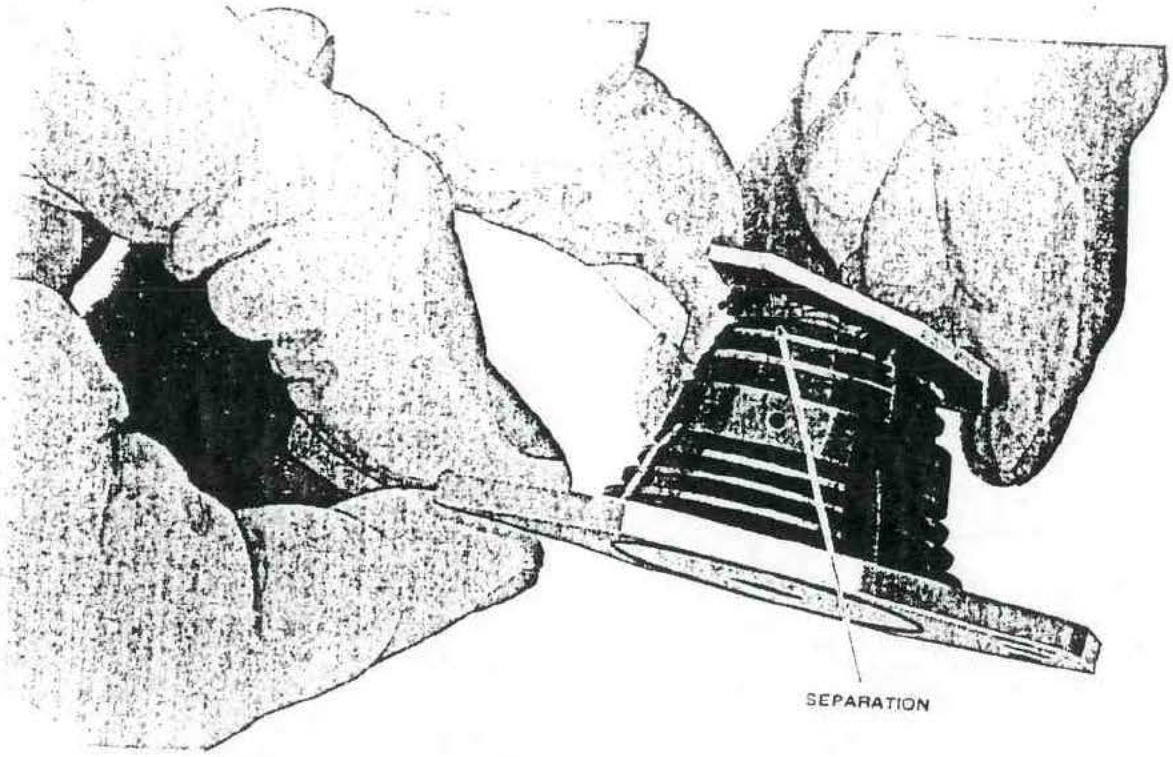
B. Painting.

CAUTION: PAINT SCHEMES (BOTH COLOR AND TYPE) USED ON THE MAIN AND TAIL ROTOR BLADES ARE FAA APPROVED. NO ALTERNATE PAINT SCHEMES ARE AUTHORIZED WITHOUT PRIOR FAA AND SIKORSKY AIRCRAFT APPROVAL.

(1) Painting of extensive areas or entire tail rotor blade is not permitted, except by direction of Sikorsky Aircraft. Blades are balanced after being painted. Any unauthorized attempt to repaint blades may change their balance and cause undesirable flight characteristics.

(2) Touchup by spot-spraying is permitted per instructions in the S-76 COMPOSITE MATERIALS MANUAL, SA 4047-76-5. These instructions include necessary precautions for cleaning and preparation of the area for application of paint. Touchup by spot spraying is used on repaired areas, and on nicked, scuffed, and scraped areas, not deep enough to require fiber glass repair.

Pages 202A and 202B deleted.



76A_55451 (B)

Snubber - Separation of Small Endplate (None Acceptable)
Figure 202

C. Ice Removal. ~~Procedures and materials for removing ice from tail rotor blades are the same as for main rotor blades.~~ (Refer to Cleaning/Painting/Ice Removal, 65-11-00.)

4. Approved Repairs Tail Rotor Blade.

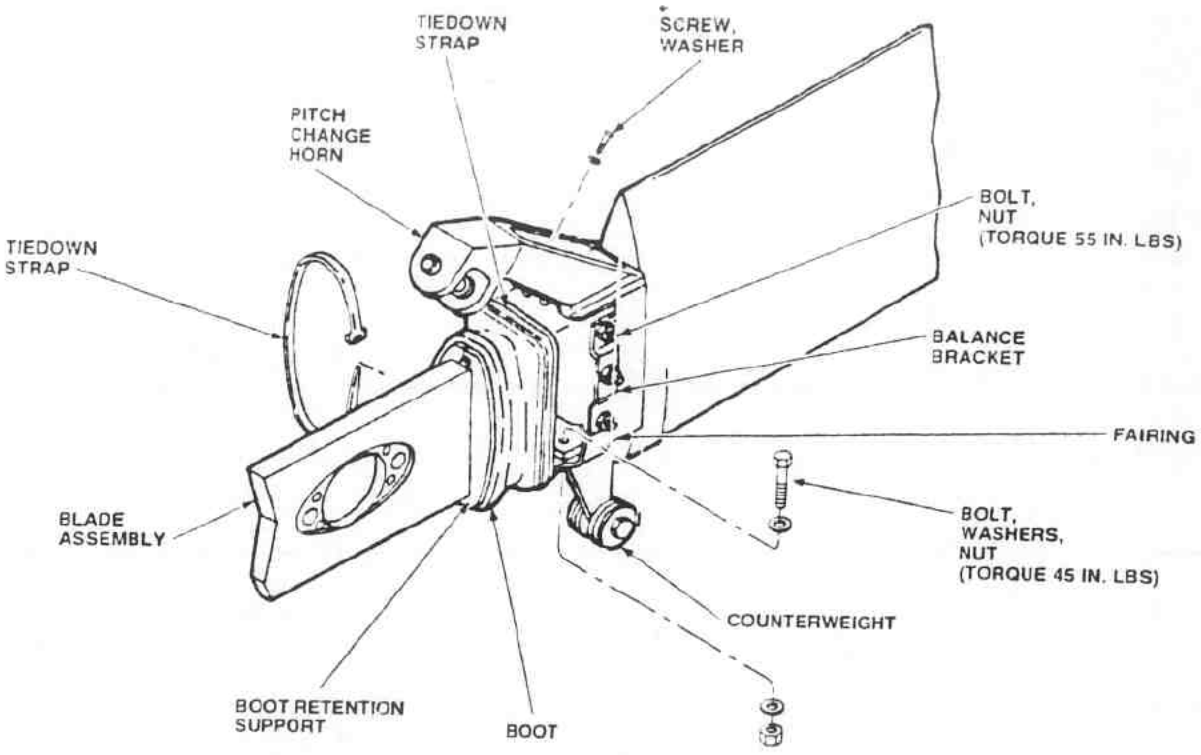
A. General. The only approved repairs for the tail rotor blades in this manual are replacement of the boot, fairing, and pivot bearings. For repair limits and other repair procedures, refer to S-76 COMPOSITE MATERIALS MANUAL, SA 4047-76-5.

B. Replacement Boot (Figure 203).

(1) Cut tiedown straps securing boot to tail rotor blade.

(2) If tail rotor blades are installed on helicopter, disconnect pitch control rod from pitch change horn by removing bolt, washers, and nut.

(3) Break seal between boot and boot retention support and stretch boot over horn and counterweight.



Tail Rotor Blade Boot and Fairing - Replacement
Figure 203

[redacted]4) Lubricate trailing edge and leading edge of blade with water and slide boot off blade.

([redacted]5) Lubricate trailing edge and leading edge of blade with water and slide new boot onto blade. Make sure egg-shaped end of boot goes on first.

[redacted]6) Slide boot along blade assembly to pitch change horn and stretch over horn and counterweight.

[redacted]) Apply sealing compound, Specification MIL-S-8802, Class B-2, to boot retention support and mating surface of boot, and secure boot to support with tiedown strap. Position tiedown strap so that strap lock is at inboard aft trailing edge of boot after tightening. Tighten tiedown strap by applying 10 - 20 pounds tensile load to its end, using adjustable hand tool, WT-197, or equivalent.

[redacted] NOTE: Make sure only black weather-resistant tiedown straps, MS3367-6-0, are installed.

NOTE: If tail rotor blades are not installed on helicopter, do not secure boot to boot retention support until after blades are installed and bolts securing boot retention support to outboard retaining plate and tail gear box output flange are installed.

8) Position outboard end of boot over flare on fiberglass fairing and secure with tiedown strap. Position tiedown strap so that strap lock is at outboard forward corner of boot after tightening. Tighten tiedown strap by applying 10 - 20 pounds tensile load to its end, using adjustable hand tool, WT-197, or equivalent.

NOTE: Make sure only black weather-resistant tiedown straps, MS3367-6-0, are installed.

) If tail rotor blades are installed on helicopter, connect pitch control rod to pitch change horn and secure with bolt (bolthead facing direction of rotation), washers, and nut, wet with zinc chromate primer, Fed. Spec TT-P-1757. Torque nut to 30 - 50 inch-pounds and install cotter pin.

NOTE: Use AN960-416L washers under nut, as required, to locate cotter pin in slot of nut.

C. Replacement Fairing (Figure 203).

1) Remove boot from tail rotor blade. (Refer to Boot Replacement.)

) Remove bolts, washers, and nuts securing fairing to pitch change horn.

) Remove screws and washers securing tab on fairing to pitch change horn.

Loosen nuts securing balance brackets and pivot bearings to blade.

) Break seal between fairing and pitch change horn using plastic scraper, and slide fairing inboard along spar. Spread fairing at split enough to remove it from spar.

5) Spread new fairing at slit enough to clear spar and install with bolt cutouts facing outboard.

7) Slide fairing outboard along spar onto pitch change horn enough to line up holes in fairing with holes in pitch change horn.

8) Apply sealing compound, Specification MIL-S-8802, Class B, to mating surfaces of fairing and pitch change horn.

NOTE: Make sure sealing compound does not spread and cover outside radius where arm joins horn box area. Outside radius must be exposed for future inspections.

[redacted]) Torque nuts securing balance brackets and pivot bearings to 55 inch-pounds.

[redacted]10) Install bolts, washers, and nuts wet with zinc chromate primer, Fed. Spec TT-P-1757, securing fairing to pitch change horn assembly. Torque nuts to 45 inch-pounds.

[redacted]1) Install screws and washers wet with zinc chromate primer, Fed. Spec TT-P-1757, securing tab on fairing to pitch change horn.

[redacted]2) Install boot on tail rotor blade. (Refer to Boot Replacement.)

D. Replacement Pivot Bearing (Figure 204).

NOTE: Replacement instructions are the same for all pivot bearings. If any pivot bearing needs replacement, the pivot bearing next to it must also be replaced.

[redacted]) Cut tiedown straps securing boot to boot retention support and fairing.

[redacted]) Break seal between boot and boot retention support and fairing, and pull boot back inboard along spar.

[redacted]) Remove bolts, washers, and nuts, and screws and washers securing fairing to pitch change horn.

[redacted]) Remove nuts securing balance bracket (if installed), fairing, and pivot bearing to spar. Remove balance bracket.

NOTE: Tag location of bracket for reinstallation in same location.

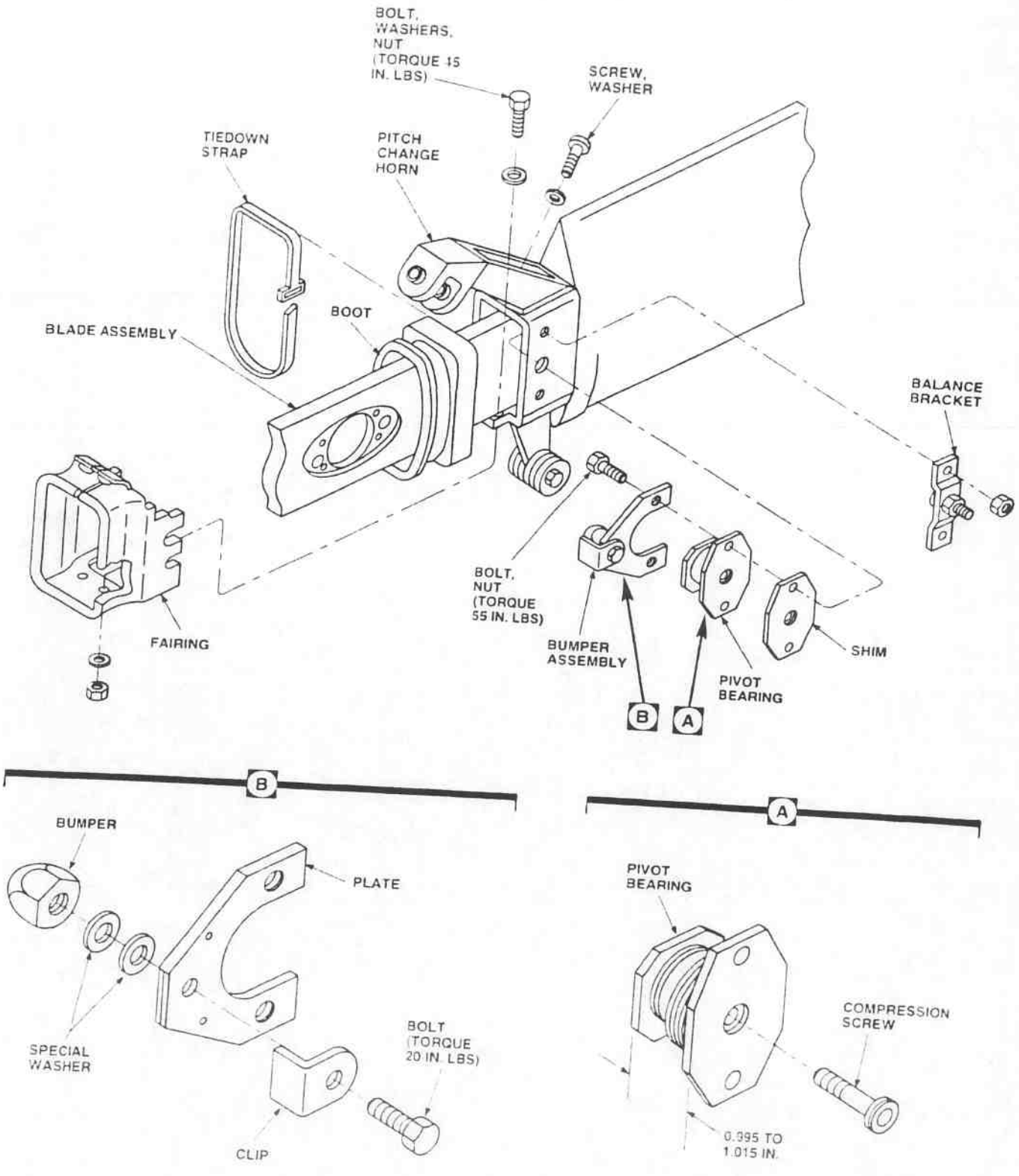
[redacted]6) Break seal between fairing and pitch change horn using plastic scraper, and slide fairing inboard along spar. Spread fairing at split and remove it from spar.

[redacted]) Remove bolts securing pivot bearing and bumper assembly to blade horn.

[redacted]) Using Allen wrench, remove ground support compression screw (CR Industries P/N B40-926377, 7/8 inch long grip) from new pivot bearing. Install ground support compression screw in old bearing. Turn ground support compression screw clockwise and compress bearing enough to remove it. Carefully remove pivot bearing, shim, and bumper assembly by pulling straight out.

[redacted]

Page 206A deleted.



CA0078

Tail Rotor Blade Pivot Bearing - Replacement
Figure 201

NOTE: Tag location of pivot bearing, shim, and bumper assembly for reinstallation in same location, if parts are to be reinstalled.

[redacted]) Inspect surface of pivot bearing retainer for evidence of indentation caused by the use of an over length ground support compression screw. If indentation is found check for retainer disbond.

[redacted]) If pivot bearing is to be reinstalled, inspect per Elastomeric Bearing (Tail Rotor Snubber) Inspection Procedures, paragraph 2.D. Inspect for bumper wear at contact point. Maximum flat spot allowed at contact point is 0.125 in. diameter.

[redacted]0) If necessary, unbolt and remove clip, special washers, and bumper from plate.

NOTE: Tag location of bumper, plate, and special washers for reinstallation in same location.

[redacted]1) Remove old sealing compound from boot, fairing, and pitch change horn using plastic scraper.

[redacted]2) If removed, coat mating surfaces of bumper assembly components with wet zinc chromate primer, Fed. Spec TT-P-1757 or epoxy primer, MIL-P-23377, and install bumper, special washers, and clip on plate and secure with bolt. Torque bolt to 20 inch-pounds and lockwire to plate. If new bumper is installed, take measurements and install required number of special washers per instructions contained in Figure 205.

[redacted]3) If new pivot bearing is being installed, take measurements and select proper shim thickness per Figure 205.

[redacted]14) Remove ground support compression screw from old pivot bearing and reinstall it in new pivot bearing. Compress pivot bearing to a height of 0.995 - 1.015 inch by turning ground support compression screw clockwise.

CAUTION: DO NOT APPLY PRIMER TO RUBBER PORTIONS OF PIVOT BEARING AND BUMPER.

[redacted]15) Coat mating surfaces of pivot bearing, shim, bumper assembly, and bolts with wet zinc chromate primer, Fed. Spec TT-P-1757 or epoxy primer, MIL-P-23377, and install pivot bearing, shim, and bumper assembly in pitch change horn.

(a) The preferred coating for mating surfaces of pivot bearing retainer and pivot bearing plate, is Mastinox 6856KD150-2 (International Celomer) or JC5A (DeSoto Titanine), because it will make future pivot bearing removal easier.

[redacted]

(b) Line up boltholes in bearing with holes in horn and install bolts.

(c) Coat exposed surfaces of pivot bearing, shim, bumper assembly, bearing retainer, and bolts with zinc chromate primer or epoxy primer. Do not use Mastinox or JC5A in this application, as it will migrate or be washed off.

NOTE: Make sure shim is completely covered with primer.

NOTE: Make sure pivot bearing and shim are installed in location noted during removal or per Figure 205.

CAUTION: GROUND SUPPORT COMPRESSION SCREW IS NOT FLIGHT HARDWARE. MAKE SURE COMPRESSION SCREW IS REMOVED BEFORE COMPLETING MAINTENANCE.

(16) Expand pivot bearing by turning ground support compression screw counterclockwise until screw is removed. Coat mating surfaces of balance bracket (or washers) and nuts with wet zinc chromate primer, Fed. Spec TT-P-1757 or epoxy primer, MIL-P-23377, and install balance bracket (if used) or washers and nuts. Torque nuts to 55 inch-pounds.

NOTE: If balance bracket is not installed, place washers under nuts.

(17) Measure installed gap between nylon bumper and bumper pad on spar, both inboard and outboard sides. Gap must be 0.100 ± 0.010 inch. If incorrect adjust per step (12).

(18) Spread fairing at split enough to clear spar and install with bolt cutouts facing outboard.

(19) Loosen nuts and slide fairing outboard along spar onto horn with cutouts under nuts. Apply sealing compound, MIL-S-8802, Class B, to mating surfaces of fairing and horn. Torque nuts to 55 inch-pounds.

NOTE: Make sure sealing compound does not spread and cover outside radius where arm joins horn box area. Outside radius must be exposed for future inspections.

CAUTION: HEAD OF BOLT MUST BE TOWARD INSIDE OF HORN.

(20) Install screws and washers, and bolts, washers, and nuts wet with zinc chromate primer, Fed. Spec TT-P-1757 or epoxy primer, MIL-P-23377, securing fairing to horn. Torque nuts to 45 inch-pounds.

(21) Apply sealing compound, Specification MIL-S-8802, Class B, to boot retention support and mating surface of boot. Pull boot back into position and secure to support and fairing with tiedown straps. Position tiedown strap at boot retention support so that strap lock is at inboard aft trailing edge of boot after tightening. Position tiedown strap at fairing so that strap lock is at outboard forward corner of boot after tightening. Tighten each tiedown strap by applying 10 - 20 pounds tensile load to its end using adjustable hand tool, WT-197, or equivalent.

[redacted]
3-912

NOTE: Make sure only black weather-resistant tiedown straps, MS3367-6-0, are installed.

Sikorsky
A United Technologies Company

Sikorsky Aircraft Corporation
6900 Main Street
P.O. Box 9729
Stratford, Connecticut 06615-9129
(203) 386-4000

ALERT SERVICE BULLETIN

No. 76-05-60

Subject ROTORS - Tail Rotor Blades - One-Time Borescope Inspection of

1 PLANNING INFORMATION

A. Effectivity

NOTE Tail Rotor Blades with less than 500 flight hours, will perform the borescope inspection to the new inspection interval published in Temporary Revisions issued in Chapter 5 of Airworthiness Limitations and Inspection Requirements Manual

All S-76 tail rotor blades in service with more than 500 flight hours Time Since New (TSN).

B Purpose

To perform a one-time borescope inspection of the leading and trailing edges of the tail rotor blade spar for cracks and delaminations through the blade thickness and for flatwise centerline cracks. This one-time borescope inspection is to be done on painted and Non-painted tail rotor blade spars.

Background
An expansion of the inspection zone on the tail rotor blade spar is required based on field experience. In addition, the inspection interval has been moved from the 1500 hour interval (Temporary Revisions will be issued alone with this ASB) to the 500 hour inspection.

C. Description

Helicopter is prepared for maintenance. This borescope inspection can be performed on installed tail rotor blades. One tail rotor blade is rotated to a convenient position. The tie-down strap securing the boot to the fairing is cut. Boot is pulled away from fairing to

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ROTORS - Tail Rotor Blades - One-Time Borescope Inspection of

1. PLANNING INFORMATION (Continued)

expose spar. Using a borescope the leading and trailing edges of the spar are checked for cracks and delaminations through the thickness. If no cracks and/or delaminations are found, reinstall boot with new tiedown strap. Perform a tail rotor balance check. Helicopter is returned to service. If cracks and/or delaminations are found, tail rotor blade is removed from service. A replacement tail rotor blade assembly is installed. Perform a tail rotor balance check. Helicopter is returned to service.

D. Compliance

Compliance is essential. The work outlined herein is to be done as follows:

(1) For tail rotor blades installed on helicopters:

(a) Perform a one-time borescope inspection identified in Section 3., within the next 500 flight hours or by December 31, 2005, whichever occurs first.

(2) For spare tail rotor blades not installed on helicopters with more than 500 flight hours TSN:

(a) Perform a one-time borescope inspection identified in Section 3., within 30 days from the issue date of this ASB.

E. Approval

Inspection Item

F. Manpower (Estimated)

	No. of Men	No. of Hours	Man-Hours
Inspect tail rotor blades	1	4.0	4.0
Remove tail rotor blades	1	1.5	1.5
Install tail rotor blades	1	1.5	1.5
Perform a tail rotor balance check	2	10.0	20.0

NOTE: Estimate does not include time required to prepare helicopter or return it to flight status.

G. Material

None

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ROTORS - Tail Rotor Blades - One-Time Borescope Inspection of

1 PLANNING INFORMATION (Continued)

H Tooling

New Part No.	Qty.	Key Word	Instructions/Disposition
A/R	1	Nonmetallic Scraper	Obtain from local supply

I Weight and Balance

None

J Electrical Load Data

None

K References

Maintenance Manual

L Publications Affected

Temporary Revisions will be issued at a later date.

SA4047-76-2-1 Airworthiness Limitations and Inspection Requirements.

SA4047-76B-2-1 Airworthiness Limitations and Inspection Requirements.

SA4047-76C-2-1 Airworthiness Limitations and Inspection Requirements.

SA4047-76-2-1 Maintenance Manual.

SA4047-76B-2-1 Maintenance Manual.

SA4047-76C-2-1 Maintenance Manual.

SA4047-76-6 Composites Materials Manual.

2 MATERIAL INFORMATION

A Basis for Material Data

Per helicopter

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ROTORS - Tail Rotor Blades - One Time Borescope Inspection of

2. MATERIAL INFORMATION (Continued)

B. Bill of Material

New Part No.	Qty	Key Word	Old Part No.	Instructions/Disposition
MS3367-6-0	4	Tiedown Strap	Same	(1)

(1) Obtain from Local Supply or contact HSI.

C. Consumable Material

WARNING

OBSERVE ALL CAUTIONS AND WARNINGS ON CONTAINERS WHEN USING CONSUMABLES. WHEN APPLICABLE, WEAR NECESSARY PROTECTIVE GEAR DURING HANDLING AND USE. IF A CONSUMABLE IS FLAMMABLE OR EXPLOSIVE, MAKE CERTAIN CONSUMABLE AND ITS VAPORS ARE KEPT AWAY FROM HEAT, SPARK AND FLAME. MAKE CERTAIN FIREFIGHTING EQUIPMENT IS READILY AVAILABLE PRIOR TO USE. FOR ADDITIONAL INFORMATION ON TOXICITY, FLASHPOINT, AND FLAMMABILITY OF CHEMICALS, CONSULT YOUR MEDICAL PEOPLE, OR THE MANUFACTURER OF THE CONSUMABLE.

A/R	Sealant	AMS-S-8802A, Class B-2 or equivalent

(Manufactured by Products Research and Chemical Corporation, 5454 San Fernando Road, Glendale, CA 91203)

As Req'd	Solvent	Isopropyl Alcohol, TT-I-735, or equivalent

(Manufactured by Aaper Alcohol and Chemical Co., 11 Isaac Shelby Dr., P.O. Box 339, Shelbyville, KY 40066-0339)

3. ACCOMPLISHMENT INSTRUCTIONS

A. Perform a one-time borescope inspection as follows. For spare tail rotor blades, proceed to section 3.A. step (2).

(1) Prepare helicopter for maintenance.

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ROTORS - Tail Rotor Blades - One-Time Borescope Inspection of

3. ACCOMPLISHMENT INSTRUCTIONS (Continued)

NOTE: This procedure applies to all four paddles (both tail rotor blades assemblies).

(2) Rotate the tail rotor blade to a convenient position to perform the inspection. Cut the tiedown strap securing the boot to the fairing. Peel boot away from fairing to expose spar.

(3) For painted spars, proceed to section B. For unpainted spars, proceed to section C.

B. Perform a one-time borescope inspection on painted spars as follows:

1-16-12

[redacted] (1) Using a borescope, inspect all four paddles (both tail rotor blades assemblies) as follows:

[redacted] a) Examine leading and trailing edges of the spar for cracks and delaminations through the thickness.

[redacted] b) Inspect the entire surface from the nylon wrap, located at the retention plate area, along the full length (approximately 12 inches beyond snubber bearing - approximately RSTA 22.00). A reference point for painted spars is when painted area ends that's 2.0 inches beyond inspection area. Refer to example of cracks and damage on the tail rotor blade spar, Figures 1 and 2; Figure 3, for painted tail rotor blade inspection area.

[redacted] c) Check for flatwise centerline cracks just inboard and outboard of the snubber bumper contact pad.

[redacted] d) Examine all remaining accessible spar surfaces for signs of other damage.

[redacted] e) If spar cracking is suspected, complete a detailed inspection to confirm a crack or damage is found and contact Sikorsky Aircraft Corporation Customer Service Engineering with results. Refer to example of an acceptable tail rotor blade spar, Figures 4 and 5.

[redacted] f) If cracks or other damage is found, remove tail rotor blade from service. Contact Sikorsky Aircraft Corporation Customer Service Engineering immediately.

[redacted] 1 Install replacement tail rotor blade per applicable maintenance manual. Proceed to section 3.A. step (2).

[redacted] g) If no cracks or other damage is found:

1 Remove old sealant using a Nonmetallic scraper.

1-16-12 2 Clean off excess sealant. Make sure there are no foreign objects, such as excess sealant in paddle.

[redacted]

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ROTORS - Tail Rotor Blades - One-Time Borescope Inspection of

3. ACCOMPLISHMENT INSTRUCTIONS (Continued)

[redacted]

3. Reinstall boot(s) with new tiedown strap(s). Apply sealant AMS-S-8802A, Class B-2 between boot retention supports and mating surface of boots per applicable maintenance manual. Let cure for 24 hours.

(2) Perform a tail rotor balance check per applicable maintenance manual.

[redacted]

(3) Return helicopter to service.

C. Perform a one-time borescope inspection on unpainted spars.

(1) Using a borescope, inspect all four paddles (both tail rotor blades assemblies) as follows.

(a) Examine leading and trailing edges of the spar for cracks and delaminations through the thickness.

(b) Inspect the entire surface from the nylon wrap, located at the retention plate area along the full length (approximately 12 inches beyond snubber bearing - approximately RSTA 2230). If a crack is found, it will show as a darker color line on the spar surface. Refer to example of cracks and damage on the tail rotor blade spar, Figures 1 and 2.

(c) Check for flatwise centerline cracks just inboard and outboard of the snubber bumper contact pad.

(d) Examine all remaining accessible spar surfaces for signs of other damage. If cracks are suspected, complete a detailed inspection to confirm cracks, or other damage as follows.

1. Using a small brush or sponge brush, apply alcohol liberally to spar leading and trailing edges from retention plate, outboard beyond pivot bearing a minimum of ten inches. Apply alcohol just inboard and outboard of the snubber bearing. Take care not to soak the rubber with alcohol. Allow the alcohol to soak for a minimum of 15 minutes. Wipe excess alcohol away. Repeat section 3.B. step (1)(a) for each paddle.

(e) If spar cracking is suspected, complete a detailed inspection to confirm a crack or damage is found and contact Sikorsky Aircraft Corporation Customer Service Engineering with results. Refer to example of an acceptable tail rotor blade spar, Figures 4 and 5.

(f) If cracks, or other damage is found, remove tail rotor blade from service. Contact Sikorsky Aircraft Corporation Customer Service Engineering immediately.

(g) Install replacement tail rotor blade per applicable maintenance manual. Proceed to section 3.B. step (2).

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ROTORS - Tail Rotor Blades - One-Time Borescope Inspection of

3. ACCOMPLISHMENT INSTRUCTIONS (Continued)

(f) If no cracks, or other damage is found:

1 Remove old sealant using a Nonmetallic scraper.

2 Clear off excess sealant. Make sure there are no foreign objects such as excess sealant in paddle.

3 Reinstall boot(s) with new tiedown strap(s). Apply sealant AMS S-8802A, Class B-2 between boot retention supports and mating surface of boots. Let cure for 24 hours.

(2) Perform a tail rotor balance check per applicable maintenance manual.

(3) Return helicopter to service.

D. Record of Compliance

Make an appropriate logbook entry to show compliance with this Alert Service Bulletin.

Upon compliance with this Alert Service Bulletin, complete attached ALERT SERVICE BULLETIN COMPLIANCE RECORD CARD and return it to Sikorsky Aircraft Corporation.

Installation of Supplemental Polyurethane Abrasion Strip Tail Rotor Blade

Feb 05/02

4-15 INSTALLATION OF SUPPLEMENTAL POLYURETHANE ABRASION STRIP TAIL ROTOR BLADE

a. Tail Blade Leading Edge Polyurethane Tape Repair

(1) This repair is to be used for the addition of a supplemental tail rotor blade leading edge polyurethane abrasion strip to the mid-span region. This repair procedure must be applied to both ends (paddles) of the cross-beam blade assembly. In addition, tail rotor dynamic balance must be checked per Maintenance Manual procedure before first flight after rework.

NOTE

Remove all loose or torn pieces of any damaged polyurethane before Step (a).

a. Mask repair area at least ¼ inch on all sides of polyurethane patch interface area as shown in Figure 4-9.

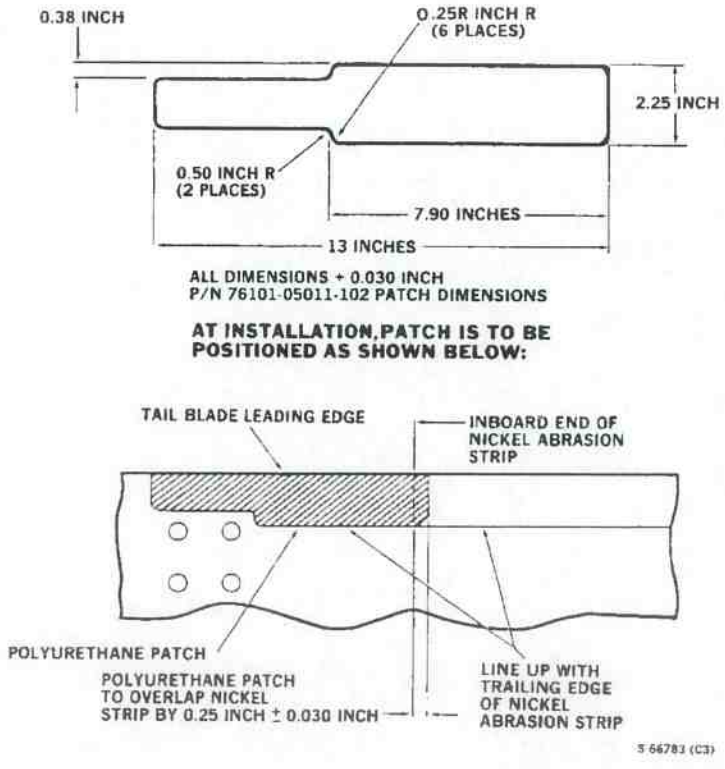

Figure 4-9. Polyurethane Patch Dimensions and Positioning

Installation of Supplemental Polyurethane Abrasion Strip Tail Rotor Blade

b. Clean enclosed area with cheesecloth dampened in methyl ethyl ketone (MEK). Repeat methyl ethyl ketone (MEK) wipe with clean cheesecloth.

c. Sand enclosed area with 150-grit paper to remove paint and scuff surface.

d. Clean sanded surface with METHYL ETHYL KETONE (MEK). Wipe dampened area with dry cheesecloth before dampness evaporates.

TECH/DATE [redacted] 3-7-12

Inspector: [redacted] **Date:** 3-7-12

e. Fill in minor dips and gouges with EA9309.3NA adhesive. Cure at 60° to 71°C (140° TO 160°F) for at least 1 hour or at 21°C to 29°C (70°F to 85°F) for at least 24 hours. Sand smooth to contour using 150-grit sandpaper. Repeat Step (d.)

f. Obtain polyurethane repair patch by procuring precut patch from Sikorsky Aircraft as P/N 76101-05011-102, or by cutting from a section of polyurethane sheet (item 36, Table 1-2) to the dimension shown in Figure 4-9. Select shiny side of polyurethane patch and clean this face by rubbing with clean, low-lint cloth moistened with METHYL ETHYL KETONE (MEK).

NOTE

Patch may have roughened, sueded finish as alternate to shiny finish.

g. Following METHYL ETHYL KETONE (MEK) wipe of face of polyurethane patch, scuff this face with 150-grit sandpaper to roughen surfaces.

h. Clean surface of the polyurethane patch with METHYL ETHYL KETONE (MEK), wipe dampened area with dry cheesecloth before dampness evaporates.

i. Brush area of blade surface and polyurethane patch with a light coating of EA9309.3NA adhesive. Apply the adhesive smoothly to both surfaces.

TECH/DATE [redacted] 3-7-12

Lot #: THOMA3648 **P.O. #:** 109987RB **Expiration Date:** 4-15-12

j. Center the polyurethane strip over the leading edge of the blade. Lightly press the polyurethane strip down on both sides of the blade, avoiding entrapment of air, until fully positioned.

Installation of Supplemental Polyurethane Abrasion Strip Tail Rotor Blade

k. Tape polyurethane strip in place with mylar tape (or equivalent) as necessary to prevent shifting.

TECH/DATE [redacted] 2-7-12

Inspector: [redacted] **Date:** 2-7-12

(l) Vacuum bag per Paragraph 4-18.

NOTE

As an alternate, two inch wide 3M polyester tape may be used to provide contact pressure as follows:
Starting at the center of the polyurethane strip, use two-inch wide polyester tape to draw the polyurethane strip down to the blade using a pressure of approximately 5 psi.
Alternately apply two-inch strips of tape on either side of the first strip (using 5 psi pressure) until the entire polyurethane strip is taped down to the blade.

m. Allow the assembly to cure at 60° to 71°C (140° to 160°F) for at least 2 hours or at 21° c to 29° c (70° to 85° F) for at least 24 hours.

n. After cure, remove tape, etc., and sand edges of excess adhesive with 150- grit sandpaper to fair the adhesive to the blade contour. METHYL ETHYL KETONE (MEK) wipe surface after sanding.

o. Repeat procedure on opposing end of tail blade assembly.

TECH/DATE [redacted] 2-7-12

Inspector: [redacted] **Date:** 2-7-12

p. Balance tail rotor per Maintenance Manual procedure.

Paint Touchup Procedure

Feb 08/02

4-10 PAINT TOUCHUP PROCEDURE

CAUTION

Paint schemes, both color type used on the main and tail rotor blades, are part of the aircraft type design approved by the FAA. Any paint scheme modifications must be approved by Sikorsky Aircraft.

CAUTION

Aerosol spray enamel is not recommended for composite surfaces because there is a potential for material strength reduction from solar radiation.

a. Preparing the area to be painted.

CAUTION

Do not allow lacquer thinner to seep into bonded joints. Thinner will weaken the bond. Do not at any time leave cloths wet with thinner lying on the blade.

1 Dampen cheesecloth with lacquer thinner, TT-T-266, or equivalent, and clean repaired area.

2 Using 360 grit abrasive paper P-P-121/P-P-101, lightly sand surface of fiberglass skins to be painted.

CAUTION

Too much sanding will result in seriously weakening of blade skin.

3 Clean area to be painted using cheesecloth dampened with METHYL ETHYL KETONE (MEK) lacquer thinner or equivalent. Wipe area with a clean cloth before dampness evaporates.

[redacted]

3-15-12

b. Applying Paint - General

NOTE

Epoxy primer (item 1 Table 1-2) is a two component kit, Mix in the proportions specified by the manufacturer.

NOTE

Epoxy primer (Item 1A, table 1-2) may be used as an alternate to (Item 1)

1 Spray/Brush on a light coat of epoxy primer (item 1 or 1A. Table 1-2).

2 Allow primer to air-dry for 1 hour.

[redacted]

Paint Touchup Procedure

Feb 08/02

3 Spray on a light coat of urethane coating (item 4 or 5 Table 1-2), or alternate paint either epoxy-polyimide (item 6, table 1-2) or acrylic lacquer (item 7, table 1-2 or polyurethane (item 7A, table 1-2). Match original color and finish (glossy/flat) of repaired area.

4 Allow urethane coating or an alternate paint to air-dry for 1 hour before handling.

c. Applying Paint – Titanium Sheath.

NOTE

A commercially available aerosol spray enamel may be applied to cover scratches, bare areas, repairs for temporary touchup. However, this should be removed and a full paint system as described herein should be applied as quickly as possible.

CAUTION

Aerosol spray enamel is not recommended for composite surfaces because there is a potential for material strength reduction from solar radiation.

1 Prepare area to be painted per paragraph 4.10.a.(1) through 4.10.a.(3).

2 Spray on one coat of wash primer (Item 8, Table 1-2).

3 Allow primer to air-dry for 1 – 4 hours.

4 Spray on a light coat of epoxy primer (Item 1 or 1A, table 1-2).

5 Allow primer to air-dry for 1 hour.

6 Spry on a light coat of urethane coating (item 4 or 5 Table 1-2), or alternate paint either epoxy-polyimide (item 6, table 1-2) or acrylic lacquer (item 7, table 1-2 or polyurethane (item 7A, table 1-2).

[redacted]

3-15-12

7 Allow urethane coating or an alternate paint to air-dry for 1 hour before handling.

d. Applying Paint – Tail Rotor Tip End, Yellow

1 Prepare area to be painted per paragraph 4-10, Steps 1. (1) through a. (3).

2 Mix and apply epoxy primer to area per note and paragraph 4-10. Steps b. (1) and b. (2) respectively.

3 Spray on a light coat of yellow urethane coating (Item 9, table 1-2).

4 Allow urethane coating to air-dry for 1 hour before handling.

[redacted]

Paint Touchup Procedure

Bell Helicopter Textron Inc. Broussard
580 St. Etienne Road, Broussard, LA 70518
Telephone: [redacted]
Telefax: [redacted]
Email: [redacted]

To: ERA Helicopters
Attn: **Michael K. Trahan**
E- [redacted]
Ph: [redacted]
Fax: [redacted]

Date: Wednesday, January 18, 2012

Sub: Quote

We at Bell Helicopter Broussard appreciate the opportunity to do business with your company and we hope to continue to do business with you in the future. If this quote is acceptable please sign below and fax to [redacted] Our standard turn around time is 60 working day on all repairs from time of quote approval. We offer a 3 year, 2000 hour prorated warranty on all repairs completed.
(AOG blades receive our highest priority and will be returned ASAP)

The quote for:

P/N: 76101-05101-041 Tail Rotor Blade
S/N: A137-00708
P/O #: RO245-011863-2012 RO#:
T/T: 14800.5

Remove and Replace Polyurethane Supplemental Strips on A & B Paddles	$	[redacted]
Install Rubber Boots on A & B Paddles	$	
Comply with 500 and 1500 Hr. Inspection	$	
Minor Repairs, Refinish and Balance	$	
TOTAL	$	

[redacted]
Approval Signature

Michael K Trahan
Print Name

2/9/12
Date

Please sign, date, and fax back to the number above ASAP.

Regards,

[redacted]

Corinne Frederick
Sr. Customer Support Specialist

This quote is good for 30 days from original date of issue.
If work is delayed due to parts availability or customer delay this quote is subject to revision.

Bell Helicopter Textron Inc. Broussard
580 St. Etienne Road, Broussard, LA 70518
Telephone: [redacted]
Telefax: [redacted]
Email: [redacted]

PACKING LIST

Date: **Friday, March 23, 2012**

From: **Bell Helicopter Textron Inc. Broussard**
580 St. Etienne Road
Broussard, LA 70518

To: **ERA Helicopters LLC**
600 Airport Service Road
Lake Charles LA 70605
Attn: [redacted]

Shipping Information

Qty	Description	Part #	Serial #	RO #
1	Tail Rotor Blade	76101-05101-042	A137-00844	RO245-011864-2012
1	Tail Rotor Blade	76101-05101-041	A137-00708X	RO245-011863-2012

Shipped – Customer Picked Up

Received By: _____ **Date:** _____

Commerce Department – 15 CFR 758.6:
"These commodities, technology or software were exported from the United States in accordance with the Export Administration Regulations. Diversion contrary to U.S. law prohibited."

1. Approving National Aviation Authority/Country:	2.	3. Form Tracking Number:
FAA/United States	**AUTHORIZED RELEASE CERTIFICATE** FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	S76T-137-00708

4. Organization Name and Address:	5. Work Order/Contract/Invoice Number:
Bell Helicopter Textron Inc. 580 St. Etienne Rd Suite A, Broussard, LA 70518	S76T-137-00708

6. Item:	7. Description:	8. Part Number:	9. Eligibility: *	10. Quantity:	11. Serial/Batch Number:	12. Status/Work:
1	Tail Rotor Blade	76101-05101-041	N/A	1	A137-00708X	REPAIRED

13. Remarks: Inspected I.A.W. CMM-SA-4047-76-5 Table 2-2, Installed boot on "A" & "B" paddles per MM-SA-4047-76-2 (65-21-01) page 203, Replaced polyurethane supplemental strips on "A" & "B" paddles per CMM-SA-4047-76-5 proc. 10 (4-16), Complied with ASB 76-65-60 (spar inspection 500 hour). Complied with 1500 hr inspection per SA-4047-76-2-1 checklist zone 3, item 4-4A, Refinished per CMM-SA-4047-76-5 proc. 6 (4-10). Balanced tail rotor component. All actions performed per Sikorsky and BHTI specs. Reference W.O. # S76T-137-00708.

T/T 14800.5

BHTI certifies that the work specified in block 12/13 was carried out in accordance with EASA Part 145 and in respect to that work the component is considered ready for release to service under EASA Part 145 Approval Number "EASA.145.6022"

14. Certifies the items identified above were manufactured in conformity to:	19. ☒ 14 CFR 43.9 Return to Service ☒ Other regulation specified in Block 13
☐ Approved design data and are in a condition for safe operation. ☐ Non-approved design data specified in Block 13.	Certifies that unless otherwise specified in Block 13, the work identified in Block 12 and described in Block 13 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

15. Authorized Signature:	16. Approval/Authorization No.:	20. Authorized Signature: [redacted]	21. Approval/Certificate No.: [redacted]
17. Name (Typed or Printed):	18. Date (m/d/y):	22. Name (Typed or Printed): Armando Martinez	23. Date (m/d/y): MAR/22/2012

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the part/component/assembly.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts parts/components/assemblies from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 14 and 19 do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

1. Approving National Aviation Authority/Country:	2.	3. Form Tracking Number:
UNITED STATES	**AUTHORIZED RELEASE CERTIFICATE** **FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG**	36671

4. Organization Name and Address:	5. Work Order/Contract/Invoice Number:
Composite Technology INC. 1001 Ave. R, Grand Prairie, TX 75050 FAA Certified Repair Station Number RW1R467K	RO245-000982-2007

6. Item:	7. Description:	8. Part Number:	9. Eligibility:*	10. Quantity:	11. Serial/Batch Number:	12. Status/Work:
1	Tail Rotor Blade	76101-05101-041	N/A	1	A137-00708X	REPAIRED

13. Remarks: X-rayed per MIL-STD-453, ultrasonic inspected per QATI 3098, 3261 & 3291, performed all scheduled inspections up to and including 1500 hr. inspection in accordance with composite MM 4047-76-5 table 3-2 and MM 4047-76-2, 65-21-00 and 65-21-01, complied with AD 94-14-20 R-1 & 89-07-12, ASB 76-65-40A, 76-65-41, 76-65-43B, 76-65-46 & 76-65-60; Performed 4000 hour torque check per RS-022N-III; Replaced both pitch horns per RS-040C-I; Replaced both primary poly strips per RS-018M-1; Replaced both secondary poly strips per SA 4047-76-5 section 4-16; Replaced laminar pad on spar of paddle B on outboard & pylon side per RS-022L-IV; Repaired skin damage on paddle B pylon side at sta. 9.0 to sta. 10.25 per OREI 2007-29; Replaced and reassembled center hardware, 2 ea fairings, 4 ea pivot bearings & 4 ea spandwise brackets per MM SA 4047-76-2 section 65-21-01; Refinished paint per drwg. 76101-05020; Static balanced per drwg. 76101-11528 amdt. 20 and drwg. 76101-11529 @ T.T. 12222.3
Full Details on W.O. 36671.

14. Certifies the items identified above were manufactured in conformity to:	19. ☒ 14 CFR 43.9 Return to Service ☐ Other regulation specified in Block 13
☐ Approved design data and are in condition for safe operation. ☐ Non-approved design data specified in Block 13.	Certifies that unless otherwise specified in block 13, the work identified in Block 12 and described in Block 13 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

15. Authorized Signature:	16. Approval/Authorization No.:	20. Authorized Signature: [redacted]	21. Approval/Certificate No.: [redacted]
17. Name (Typed or Printed):	18. Date:	22. Name (Typed or Printed): **Linda J. Gray**	23. Date (m/d/y): **10-25-07**

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the part/component/assembly.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness accepts parts/components/assemblies from the airworthiness authority of the country specified in Block1.

Statements in Blocks 14 and 19 do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

ERA HELICOPTERS
ERA 50C
OCT 2 9 2007
RECEIVING INSP.
X 000982

FAA Form 8130-3 (6-01) *Installer must cross-check eligibility with applicable technical data. NSN: 0052-00-012-9005

11-10-10

Era HELICOPTERS LLC a SEACOR company

600 Airport Service Road
Lake Charles LA 70605

Form M-003

REPAIRABLE

Part Description:	Part Number:	Serial Number:	Comp. ID:
TAIL ROTOR BLADE	76101-05101-041	A137-00708X	ROT-853

TSN:	TSO:	Next Maintenance:	Next Maintenance Due:
14800.5	14793.3	S76 TRB SPAR PLUG INSP	14818FH

Object Type:	Ref. Doc. No:	Date:	PO Number:
Tracked Component	TL-042657-2012	01/07/2012	RO245-000982-2007

****REMOVED REPAIRABLE DETAILS****

A/C Registration Number:	NHA Part Number:	NHA Serial Number:	Position Code:
N578EH			T/R BLADE ASSEMBLY 1
Component Replacement No:	Removed By:	Removal Station:	Removal Date:
TLCR-010700-2012	Hudson, David W.	LAKE CHARLES	01/07/2012
Removal Code:	Reason for Removal Remarks:		
EROSION	1500 hr inspection due, paint erroded		

Comments:

[X] Era Owned Part [ ] Vendor Owned Part [ ] Warranty Core Warranty Claim No:

Date Generated: 01/07/2012

Era HELICOPTERS LLC a SEACOR company

600 Airport Service Road
Lake Charles LA 70605

Form M-003

REPAIRABLE

Part Description:	Part Number:	Serial Number:	Comp. ID:
TAIL ROTOR BLADE	76101-05101-041	A137-00708X	ROT-853

TSN:	TSO:	Next Maintenance:	Next Maintenance Due:
14800.5	14793.3	S76 TRB SPAR PLUG INSP	14818FH

Object Type :	Ref. Doc. No :	Date:	PO Number:
Tracked Component	TL-042657-2012	01/07/2012	RO245-000982-2007

****REMOVED REPAIRABLE DETAILS****

A/C Registration Number:	NHA Part Number :	NHA Serial Number:	Position Code:
N578EH			T/R BLADE ASSEMBLY 1
Component Replacement No:	Removed By:	Removal Station:	Removal Date:
TLCR-010700-2012	Hudson, David W.	LAKE CHARLES	01/07/2012
Removal Code:	Reason for Removal Remarks:		
EROSION	1500 hr inspection due, paint erroded		

Comments:

[X] Era Owned Part [ ] Vendor Owned Part [ ] Warranty Core Warranty Claim No:

Date Generated: 01/07/2012

FAX [redacted]

ATTN: JIM DE MARR - KEYSTONE HELICOPTER

MSG. NO.: CBT-TC-91-1512 DT: 7 NOV 91

FM: ROY K.MOORE - SIKORSKY AIRCRAFT - FAX: [redacted]

RE: A/C 760274, TRB S/N A137-00708, SPAR S/N A118-01207, TSN: 4364.8 - PAINT CRACK INDICATION ON EDGE OF SPAR APPROX 3 IN. OUTBOARD OF PIVOT BEARING.

IT IS UNDERSTOOD THAT THE 2 AREAS WHERE THE PAINT CRACKS WERE DETECTED, HAD POOR PAINT ADHESION WHICH WAS IDENTIFIED BY THE PAINT BEING VERY EASILY REMOVED WITH A PLASTIC SCRAPER. THE FOLLOWING INSPECTION OF THE SPAR IS TO BE PERFORMED AS FOLLOWS:

* REMOVE PAINT FROM SPAR IN THE SURROUNDING AREA
* PERFORM ALCOHOL WIPE IN THESE AREAS FOR 1/2 HOUR
* AT THE SAME TIME, FLEX THE SPAR FOR POSSIBLE CRACK INDICATION.

IF INSPECTION OF SPAR CONFIRMS THE CRACKS TO BE IN THE PAINT ONLY, RESTORE PROTECTIVE FINISH ON SPAR AND CONTINUE T/R BLADE ASSY IN SERVICE.
NOTE: INSPECT THIS AREA OF THE SPAR AT THE NEXT SCHEDULED 500 HR INSPECTION.

REGARDS,

[redacted]

CC: L.KROHELSKI, A.SMITH, C.LYNCH, M.GAMAUF, P.POTTS, REPS, W.REINFELDER

Repair Order

Mail Invoices To :

ERA HELICOPTERS LLC
600 Airport Service Road
LA Lake Charles
USA 70605

Repair Order # : RO245-011863-2012
Amendment # : 0
THIS NUMBER MUST APPEAR ON ALL INVOICES, PACKING SLIPS, PACKAGES & CORRESPONDENCE

Supplier : ROTOR BLADES INC
580 ST ETIENNE RD, STE H
BROUSSARD, LA 70518 US

Ship To : ERA HELICOPTERS LLC
600 AIRPORT SERVICE ROAD
LAKE CHARLES, LA 70605 USA

Phone # : [redacted]
Fax # :

Supplier #	RO Date	Ship By	F.O.B	Pay Term
8100-8100-S3	01/09/2012	UPS GRND	ORIGIN	NET 30

Contact	Phone # / Ref Doc #	Quotation #	Approved By

Line #	Part # / Description	Serial # / Lot #	Due Date	Qty.	UOM	Repair Cost
1	76101-05101-041 TAIL ROTOR BLADE /	A137-00708X	02/09/2012	1	EA	

Repair Order # : RO245-011863-2012

Maintenance Information Details

Part # : 76101-05101-041 **Part Desc :** TAIL ROTOR BLADE **Serial # :** A137-00708X

Component Details				Parameter Details (FH)			
Component ID	ROT-853	Level Code		TSN	14800.5	TSI	312.1
EIPN Comp ID		Position Code		TSO	14793.3	TSR	14793.3

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
AD86-19-14	Others			32263 Days	31769 Days
Replacement of listed serial numbered tail rotor assemblies					
AD94-14-20 R1	Others			32263 Days	31769 Days
Inspection of the tail rotor blade spar elliptical centering plug for disbonding and the addition of a retaining pad on the pitch change shaft					
S76 TR BLADE INSP	Inspection	07/06/2011	14800.5 HR		187.9
Tail Rotor Blade Boroscope Inspection		14488.4 HR		500 HR	
S76 TRB SPAR PLUG INSP	Inspection		14800.5 HR		17.4
Tail Rotor Blade Spar Plug Inspection		13317.9 HR		1500 HR	

Part # : 76101-05020-045 **Part Desc :** TAIL ROTOR BLADE **Serial # :** 1819

Component Details				Parameter Details (FH)			
Component ID	ROT-4121	Level Code	1.1	TSN	5136.6	TSI	5129.4
EIPN Comp ID		Position Code	T/R BLADE 1	TSO	5129.4	TSR	5129.4

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
S76 TR BLADE RET	Retire		5136.6 HR		14863.4
Tail Rotor Blade Retirement				20000 HR	

Part # : 76101-05020-045	Part Desc : TAIL ROTOR BLADE	Serial # : 1820

Component Details				Parameter Details (FH)			
Component ID	ROT-4122	Level Code	1.2	TSN	5136.6	TSI	5129.4
EIPN Comp ID		Position Code	T/R BLADE 2	TSO	5129.4	TSR	5129.4

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
S76 TR BLADE RET	Retire		5136.6 HR		14863.4
Tail Rotor Blade Retirement				20000 HR	

Part # : 76101-05006-042	Part Desc : HORN ASSY, TR	Serial # : B361-00269

Component Details				Parameter Details (FH)			
Component ID	ROT-4123	Level Code	1.3	TSN	2578.2	TSI	2578.2
EIPN Comp ID		Position Code	T/R HORN 1	TSO	2578.2	TSR	2578.2

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
AD89-07-12.	Others		2578.2 HR		9421.8
Establish a life limit for tail rotor horn, P/N 76101-05006 of 12,000 hours time in service, remove all items that have exceeded this value.				12000 HR	
S76 TR HORN RET	Retire		2578.2 HR		9421.8
Tail Rotor Horn Retirement				12000 HR	

Part # : 76101-05006-042	Part Desc : HORN ASSY, TR	Serial # : B361-00328

Component Details				Parameter Details (FH)			
Component ID	ROT-4124	Level Code	1.4	TSN	2578.2	TSI	2578.2
EIPN Comp ID		Position Code	T/R HORN 2	TSO	2578.2	TSR	2578.2

Task Details

Work unit #	Maintenance Type	LPD	Current Value	Calender Interval	Remaining Value
Work Unit Desc	Ref Document #	LPV		Usage Interval	
AD89-07-12.	Others		2578.2 HR		9421.8
Establish a life limit for tail rotor horn, P/N 76101-05006 of 12,000 hours time in service, remove all items that have exceeded this value				12000 HR	
S76 TR HORN RET	Retire		2578.2 HR		9421.8
Tail Rotor Horn Retirement				12000 HR	

8130-3 REQUIRED

Please Inspect/Repair/Overhaul as required in accordance with Manufacturer's Manual and current Regulatory Requirements.

N578EH 1500 hr isnpection due paint erroded inspect and repair

Please email quote with turn around time to roquotes@erahelicopters.com

Legends : **F.O.B- Free On Board, QTY-Quantity, UOM-Unit Of Measurement, EIPN-End Item Part Number, TSN- Time Since New, TSI-Time Since Inspection, TSO-Time Since Overhaul, TSR-Time Since Repair, LPD-Last Performed Date, LPV-Last Performed Value.**

THIS ORDER IS EXPRESSLY MADE AND ITS ACCEPTANCE IS EXPRESSLY LIMITED TO THE TERMS AND CONDITIONS CONTAINED HEREIN AND ATTACHED TO LAST PAGE		TOTAL :
	AUTHORIZED SIGNATURE	

CONSIGNEE

CONSIGNEE MEMO

ROTOR BLADES INC
SUITE A
580 ST ETIENNE RD
BROUSSARD [redacted] LA 70518

PO# NONE

UPS FREIGHT PHONE NUMBER [redacted]

DELIVERING TRAILER
262289 OVNT

FREIGHT BILL NUMBER [redacted]

CITY RTE/BYD SCAC	DEST
026N	BRG

PICK UP DATE	ORIG
01/11/12	BEA

SHIPPER [redacted]
ERA HELICOPTERS LLC

600 AIRPORT SERVICE RD
LAKE CHARLES [redacted] LA 70606 [redacted]

ADV CAR

BL# NONE

AD UF BD

COLLECT
THIS AMOUNT [redacted]

UPS Freight℠

www.upsfreight.com

UPGF [redacted]

#PCS	HM	PT	DESCRIPTION OF ARTICLES AND SPECIAL MARKINGS	WEIGHT(LBS)	NMFC	RATE	CHARGES
			1 PIECE(S) COUNTED AND VERIFIED ON				
			1 SK HANDLING UNIT(S) WITH THE FOLLOWING:				
1		CR	TAIL ROTOR BLADES	210	000070-00		
			LTL FUEL ADJUSTMENT				
			UPS WORLDSHIP				
			* * * * * ATTENTION * * * * *				
			REPAIR				
			337 839 2125 C/PH#				
			BILL TO: [redacted]				
			ERA HELICOPTERS LLC				
			UPGF [redacted]				
			UPGF 560 08/01/11 C N 16039				

1 <TTL PCS

PRINT NAME

SIGNATURE
X

TTL WT > 210

FIRM

ODOM	ARRIVE	DEPART	TOTAL CHARGES
			PPD

PIECES RCVD

WRAP INTACT?
YES NO?

RECEIVED THE ABOVE PROPERTY IN GOOD CONDITION EXCEPT AS NOTED
RECORD EXCEPTIONS & DESCRIPTIONS OF GOODS IN BODY OF FORM ABOVE

DATE

DRIVER NAME

UPS WS14.0.21

STRAIGHT BILL OF LADING- SHIPPING ORDER NOT NEGOTIABLE
UPS FREIGHT (UPGF)

WEB SITE: www.ups.com
DATE: 01/11/2012

CONSIGNEE
ROTOR BLADES INC
ATTENTION: REPAIR
UPS Freight cannot deliver to a P.O. Box
580 ST. ETIENNE ROAD # A
BROUSSARD, LA 70518
US
PHONE: [redacted]

SHIPPER
ERA HELICOPTERS LLC
ATTENTION: HOWARD MOORE

600 AIRPORT SERVICE RD
LAKE CHARLES, LA 706070610
US
PHONE: [redacted]

BILL TO
ERA HELICOPTERS LLC
ATTENTION: HOWARD MOORE

600 AIRPORT SERVICE RD
LAKE CHARLES, LA 706070610
US
PHONE: [redacted]

BILLING METHOD
- [x] Prepaid
- [ ] Collect
- [ ] Third Party

Received $..... to be delivered in the prepayment of the charges on the property described hereon. (agent or cashier) .....

☐ **GUARANTEED DELIVERY REQUESTED (if box is checked)**

By checking this box, the Payer requests UPS Freight to guarantee delivery of this shipment according to UPS Freight's transit schedule and agrees to pay 25% more ($30 minimum) above the cost normally incurred with this service. Payer will not be liable for payment if shipment fails to deliver on scheduled day.

DESCRIPTION OF ARTICLES, WEIGHT, NMFC, & CLASS ARE SUBJECT TO CORRECTION

# OF PIECES	PKG TYPE	HM *	DESCRIPTION OF ARTICLES & SPECIAL MARK	WEIGHT	<UOM>	NMFC	CLASS
1	Package		TAIL ROTOR BLADES	210	LBS		70
1			TOTALS:	210			

SHIPPED AS: 1 HANDLING UNITS AND LOOSE

*Marked with an "X" to designate Hazardous Materials as defined in Title 49 of the Code of Federal Regulation
Hazardous Material Emergency Contact Number:

Additional Services: (CHARGES MAY APPLY)
- ☐ CALL BEFORE DELIVERY
- ☐ CONSTRUCTION SITE DELIVERY
- ☐ HOLIDAY DELIVERY
- ☐ RESIDENTIAL DELIVERY
- ☐ WEEKEND DELIVERY
- ☐ INSIDE DELIVERY
- ☐ LIFT GATE REQUIRED
- ☐ HOLIDAY PICKUP
- ☐ INSIDE PICKUP
- ☐ RESIDENTIAL PICKUP
- ☐ WEEKEND PICKUP
- ☐ SORT AND SEGREGATE _____ Pieces

REFERENCE NUMBERS:

COD FEE	COD AMT		REMIT COD CASH/ CHECK TO:
☒ Prepaid ☐ Collect	$	☐ CONSIGNEE CHECK ACCEPTABLE ☐ CERTIFIED CHECK OR CASH	

CARRIER LIABILITY: Carrier liability for loss or damage will be the lesser of (1) the actual invoice of the commodities or article(s) lost, damaged, or destroyed, or * (2) the amount determined from applicable limited liability provisions of the NMFC, or (3) the limited liability as stated in the applicable governing tariffs, unless **Excess Declared Value Coverage is specifically requested along with the amount of coverage needed in writing on the bill of lading at the time of shipment and applicable charges are paid. Maximum carrier liability is limited to $25.00 per pound per package and $100,000 per shipment. Liability for commodities or articles other than new is limited to $.10 per pound per package (and up to a maximum of $2.50 per pound per package when Excess Declared Value Coverage is requested). Liability for specific commodities or articles described in the UPGF rules tariff item 168 section 5 is limited to $2.00 per pound per package. Liabilities for commodities or articles subject to exception rating (FAK) is limited as described in the tariff. Certain items may be subject to a limited declared value, with a choice of rates under the tariff. You are advised to review the applicable tariff provisions before stating a value. ***Refer to the current tariff UPGF 102 series for complete details. Where a "rate" is dependant on a released, declared or actual value of the property is hereby specifically stated by the shipper not to be exceeding _____ per _____

**Shipper requests Excess Declared Value in the amount of $_____

RECEIVED, subject to the individually determined rates or contracts that have been agreed upon in writing between the carrier and shipper, if applicable, otherwise to the rates, classifications and rules that have been established by the carrier and are available to the shipper upon request, ***Now available at www.upsfreight.com - UPGF 102 Series Rules Tariff - Electronic Bill of Lading. The property described above in apparent good order, except as noted (contents and condition of contents of packages unknown) marked, consigned, and destined as shown below, which says carrier agrees to carry to destination, if on its route, or otherwise to deliver to another carrier on the route to destination. Every service to be performed hereunder shall be subject to all the conditions not prohibited by law, whether printed or written, herein contained, including the conditions on the back hereof, which are hereby agreed to by the shipper and accepted for himself and his assigns. Where a third party bill or broker exists, carrier holds both the shipper and consignee liable for freight charges.

Subject to Section 7 Terms and Conditions, if this shipment is to be delivered to the consignee without recourse on the consignor, the consignor shall sign the following statement: UPS Freight may decline to make delivery of the shipment without payment of freight and all other lawful charges.

(Signature)_____

ODOMETER	ARRIVE	DEPART	DESTINATION	TRAILER NUMBER	LINEAR FEET OF SHIPMENT
				SEAL # APPLIED	
				BEYOND SCAC:	CROSS REF PRO #
				☐ SHIPPER LOAD/ UNLOAD	

BEA 290534506

This is to certify that the above named materials are properly classified, described, packaged, marked, and labeled and are in proper condition for transportation according to the applicable regulations of the Department of Transportation.

Firm name: Era Helicopters LLC

Signed by: [redacted]

UPS Freight shall have no liability or responsibility whatsoever in connection with this bill of lading if the shipper did not tender the shipment to UPS Freight or its agent.

Carrier: UPS Freight	Driver:
Date Received:	UPS Freight Piece Count

Bell Helicopter Textron Inc. Broussard
580 St. Etienne Road, Broussard, LA 70518
Telephone: 337-839-2119
Telefax: 337-839-2122
Email: cfrederick@bellhelicopter.textron.com

To: ERA Helicopters
Attn: **Michael K. Trahan**
E- [redacted]
Ph:
Fax:

Date: Wednesday, January 18, 2012

2nd quote Sent 2/9/12

Sub: Quote

We at Bell Helicopter Broussard appreciate the opportunity to do business with your company and we hope to continue to do business with you in the future. If this quote is acceptable please sign below and fax to [redacted] Our standard turn around time is 60 working day on all repairs from time of quote approval. We offer a 3 year, 2000 hour prorated warranty on all repairs completed.
(AOG blades receive our highest priority and will be returned ASAP)

The quote for:

P/N: 76101-05101-041 Tail Rotor Blade
S/N: A137-00708
P/O #: RO245-011863-2012 RO#:
T/T: 14800.5

Remove and Replace Polyurethane Supplemental Strips on A & B Paddles	$	[redacted]
Install Rubber Boots on A & B Paddles	$	
Comply with 500 and 1500 Hr. Inspection	$	
Minor Repairs, Refinish and Balance	$	
TOTAL	**$**	

Approval Signature *Print Name* *Date*

Please sign, date, and fax back to the number above ASAP.

Regards,

[redacted]

Corinne Frederick
Sr. Customer Support Specialist

This quote is good for 30 days from original date of issue.
If work is delayed due to parts availability or customer delay this quote is subject to revision.

Bell Helicopter Textron Inc.
Tail Rotor Blade
76101-05101-041
DISCREPANCY/QUOTE SHEET

Customer: Era Aviation

Part No: 76101-05101-041 Serial No: A137-00708x

Blade T.T: 14,800.5 Date Inspected: 1-16-2012 Inspector: [redacted]

	Repair Data: Inspected and repaired per Sikorsky Specifications Inspected I.A.W. CMM SA 4047-76-5 Table 2-2,	*Part Number*	Repair Procedure # Composite Materials	Repair Procedure # ORI	Verified Complete	Completed By
Discrepancy	Polyurethane supplemental strips eroded on A & B paddles	76101-05011-102	CMM SA-4047-76-5 proc 10 (4-16)	1162.00	✓	
Corrective Action	Removed and Replaced polyurethane supplemental strips on A & B paddles	qty 2				
Discrepancy	Rubber boot torn on A & B paddles	76101-05008-101 qty 2	MM SA 4047-76-2 (65-21-01) page 203	836.00 ✓		
Corrective Action	Installed boot on A & B paddles					
Discrepancy	Comply with 500 hr inspection		ASB 76-65-60	1848.00 ✓		
Corrective Action	Complied with (spar inspection 500 hour)					
Discrepancy	Comply with 1500 hr inspection		SA-4047-76-2-1 checklist zone 3, item 4-4A	✓		
Corrective Action	Complied with 1500 hr inspection					
Discrepancy						
Corrective Action						
Discrepancy						
Corrective Action						
Discrepancy						
Corrective Action						
Discrepancy						
Corrective Action						

Continued on Next Page

[redacted]

Bell Helicopter Textron Inc.
Tail Rotor Blade
76101-0510-041
DISCREPANCY/QUOTE SHEET

Customer:	Error! Not a valid link. Era Aviation	Serial No:	Error! Not a valid link. A137-00708x

Repair Data: Inspected and repaired per Sikorsky Specifications		*Part Number*	Repair Procedure # Composite Materials	Repair Procedure # O R I	Verified Complete	Completed By
Discrepancy		[redacted]	✓			
Corrective Action						
Discrepancy Corrective Action	☐ *Perform RII inspection (when required)* ☐ *Accomplished all applicable PHI RII Requirements*					
Discrepancy	☑ Refinish per CMM SA4047-76-5 Procedure # 6 4-10					
Corrective Action	☑ Refinished per CMM SA4047-76-5 Procedure # 6 4-10					
Discrepancy	☑ Balance					
Corrective Action	☑ Balanced					

TRACK	**MINUTES**			
BLADE SERIAL NO.	A137-00708x			
BLADE DWG. NO.	76101-0510-041			
MFG. 5-16-84	**RS**	-040C-I	**RS**	
REP.	**RS**	-018M-I	**RS**	
REP.	**RS**	-022L-IV	**RS**	
REP.	**RS**		**RS**	
REP.	**RS**		**RS**	

Frederick, Corinne

From: Frederick, Corinne
Sent: Wednesday, January 18, 2012 12:58 PM
To: [redacted]
Subject: Quotes for Agusta Tail Rotor Blades, S76 Tail Rotor and Tip Cap
Attachments: DOC.PDF

Good Afternoon,
Attached please find the quotes for the following blades... Once approved please fax or email back to my attention for work to begin... Please don't hesitate to contact me if you have any questions.
Thanks

709 Agusta Tail Rotor Blades
N-987 - Please send Original Historical Records
M-813 - Please send Original Historical Records

S76 Tip Cap
A175-04199

S76 Tail Rotor Blades
A137-00844
A137-00708

Corinne Frederick
Sr. Customer Support Specialist

Bell Helicopter Textron Inc. Broussard
580 St. Etienne Road.
Broussard, LA 70518
Phone [redacted]
Fax [redacted]
Email: [redacted]

Bell Helicopter Textron Inc.

Repair Station # R8DR713Y

CUSTOMER BLADE RECEIVER

Customer Order # CHINOOK

Work Order # 576T-137-00708

DO NOT MIX CUSTOMER SHIPMENTS USE ONE RECEIVER FOR EACH CUSTOMER AND BLADE

Customer: ERA AVIATION Date Received: JAN 12 2012 Received By: [redacted]

TYPE			Part Number	Serial Number	Time Since New	Records Yes	Records No	Date Shipped	Box Number
M/R	T/R	T/C							
	X		76104-05101-041	A137-00705Y	14800.5	X			WOODEN BOX

LOCATION OF CONTAINERS:

Condition of Container:	☐ **Good** ☐ **Fair** ☐ **Repairable**	☐ **Blade Box Needs To Be Replaced**
Condition of Contours:	☐ **Good**	☐ **Replace Contours**

To insure that Containers are returned to the proper Customer be sure to accurately mark each container with Customer Name and Box # listed on this form.

Work to be performed: STANDARD REPAIR

Comments/Special Instructions: _____

Incoming Paint Scheme: Select One _____ High Visibility Paint Scheme _____ Standard Paint Scheme

Freight in: UPS Freight out: _____ Contact Name: _____

Purchase order Information: _____ Telephone: _____ Fax: _____

When completed forward a copy to the General Manager

Customer support must be notified if the container is not suitable for shipment of repaired items. The Original completed CUSTOMER BLADE RECEIVER, and all documentation removed from the container is put into a folder and forwarded to the Quality Control Department for incoming blade inspection.

RBI –L – 001

Bell Helicopter
A Textron Company

Remittance Address
Bell Helicopter Textron Inc.
23413 Network Place
Chicago, IL 60673-1234
E-mail: blades@bellhelicopter.com

CUSTOMER ORDER ACKNO

Order Number: **C12116R**
Date: **3/26/2012**
Page: **1**

Sold To

ERA HELICOPTERS LLC
P O BOX 13038
FORT LAUDERDALE, FL 33316
United States

PHONE: [redacted] FAX: [redacted]
E-MAIL: [redacted]

Ship To

ERA HELICOPTERS LLC
600 AIRPORT SERVICE ROAD
LAKE CHARLES, LA 70605
UNITED STATES

CUSTOMER ID	CUSTOMER P.O.	PAYMENT TERMS	FREIGHT TERMS
[redacted]	[redacted]	Net 30	Freight prepaid

SALES REPRESENTITIVE	SHIPPING METHOD	F.O.B.	SHIP DATE
	CUSTOMER PICKED UP	ORIGIN	2/9/2012

QUANTITY ORD	QUANTITY SHP	QUANTITY BCK	PART ID	DESCRIPTION	UNIT PRICE	EXTENDED PRICE
1.00	0.00	0.00	TRLAMDRP	TAIL ROTOR LA 212, 412 BLAD Work Order ID: Product Code: ***DOMLABR*** Deliver by 2/9/2012 P/N 76101-05101-041 TR S/N A137-00708X TSN 14800.5 RO# RO245-011863-2012	[redacted]	

SUB TOTAL [redacted]

TOTAL ORDER AMOUNT

Bell Helicopter Textron Inc.
441 INDUSTRIAL PARK DRIVE
PINEY FLATS, TN 37686
PHONE [redacted]
FAX

PACKLIST

Packlist ID: **PL12049RB**
Date: **03/26/2012 12.0**
Page: **1**

Sold To Address	Ship To Address
ERA HELICOPTERS LLC P O BOX 13038 FORT LAUDERDALE, FL 33316 United States	ERA HELICOPTERS LLC 600 AIRPORT SERVICE ROAD LAKE CHARLES, LA 70605 UNITED STATES

CUSTOMER ID	CUSTOMER PO	PAYMENT TERMS	F.O.B.
[redacted]	[redacted]	Net 30	ORIGIN
SALES REP ID	**SHIPPING METHOD**	**SHIP DATE**	**OUR ORDER NUMBER**
	CUSTOMER PICKED UP	3/23/2012	C12116RB

QUANTITY					CUSTOMER PART NO.	OUR WORK ORDER NO.
ORD	SHP	BCK	PART ID	DESCRIPTION		
1.00	1.00	0.00	TRLAMDRP	TAIL ROTOR LA 212, 412 BLADE REP		

Ship Weight:

Shipped Unit of Measure: EA
P/N 76101-05101-041 TR
S/N A137-00708X
TSN 14800.5
RO# RO245-011863-2012

ORDER SPECIFICATIONS

BILLING CHECK LIST

Serial # - A137-00708X

PO or RO # - R0245-011863-2012

SHIPPING CHECK LIST

Domestic Shipment

☐ Called Customer to confirm Ship to and Bill to Address

Ship To: ERA Helicopters

Bill To: same

Contact Person - [redacted]
Phone Number - [redacted]

☐ Shipping Method
Carrier - Customer
Account Number - will
Shipping Type - pick up

International Shipment

☐ Approved shipment with Ruth Tapp
☐ Sent with SED

PAYMENT CHECKLIST

☐ Due on Receipt
☑ Net 30 Days
☐ COD
☐ Over Credit Limit (Must Contact Randal before Shipping)

☐ Sent in Wire Transfer (Date - _____, Amount - _____)

Credit Card Information
Type of Card - _____
Card Number - _____
Exp. Date - _____
Code - _____
Zip Code - _____
Name on Card - _____

Total amount Due - $_____ Total Shipping Charge - $_____

SIKORSKY AIRCRAFT Division of UNITED TECHNOLOGIES

COMPONENT LOG CARD

PART NO. 76101-05101-041 NOMENCLATURE TAIL ROTOR BLADE SERIAL NO. A137-00708X

651700

AIRCRAFT NO.	DATE INSTALLED	DATE REMOVED	AIRCRAFT TIME	COMPONENT TIME T.S.O.	COMPONENT TIME T.T.	REASON FOR REMOVAL
760274	30Nov.1984		NEW	NEW	NEW	
760274		2 OCT 87	2230.0	New	2230.0	C/W 76-65-40
760274	13 Nov 87		2315.9	New	2230.0	ERROR
760274		17 NOV 90	3915.0	NEW	3915.0	REPAINT
760274	29 NOV 90		3915.0		3915.0	CORRECTED
760274		6-22-01	3996.0	NEW	3995.1	INSPECT & REPAIR

[redacted]

SA 7343-9 REV A.

OVERHAUL INTERVAL

RETIREMENT TIME 12,000 HRS

COMPONENT LOG CARD

PART NO. 76101-05001-044 NOMENCLATURE BLADE ASSY;TAIL RTR OUTBD SERIAL NO. A13700708X

AIRCRAFT NO.	DATE INSTALLED	DATE REMOVED	AIRCRAFT TIME	COMPONENT TIME T.S.O.	COMPONENT TIME T.T.	REASON FOR REMOVAL
ZS-RLG	11-14-87		2316	2230		

Imperial

OVERHAUL INTERVAL ON CONDITION.

RETIREMENT TIME

...EQUIPMENT TO MAJOR COMPONENT

TAIL ROTOR BLADE	2. PART NUMBER 76101-05101-041	3. SERIAL NUMBER A137-00708

PART NAME A.	PART NUMBER B.	SERIAL NUMBER C.	DATE D.	PERFORM. ORGAN-IZATION E.	TIME SINCE NEW ON ITEM SHOWN IN BLOCK 1	TIME SINCE NEW ON LIFE LIMITED ITEMS SHOWN IN COLUMN A.		REASON FOR REMOVAL
						INSTALLED	REMOVED	
Spar Fastener	76101-05023-105	NONE	6-22-84	Sikorsky	NEW	NEW		
Spar Fastener	76101-05023-107	NONE	"	"	"	"		
Spar Fastener	76101-05023-108	NONE	"	"	"	"		
Spar Fastener	76101-05023-109	NONE	"	"	"	"		
Spar	76101-05017-045	A116-01207	"	"	"	"		
Blade	76101-05020-045	1819	"	"	"	"		
Blade	76101-05020-045	1820	"	"	"	"		
Pitch Horn	76101-05006-042	1819	"	"	"	"	12222.3	time x-pired
Pitch Horn	76101-05006-042	1820	"	"	"	"	12222.3	time x-pired
Pitch Horn	76101-05006-042	B361-00269	10-25-07	CTI	12222.3	NEW		
Pitch Horn	76101-05006-042	B361-00328 ~~00264~~	10-25-07	CTI	12222.3	NEW		

UNITED TECHNOLOGIES
SIKORSKY AIRCRAFT

AIRCRAFT MAINTENANCE LOG

T/R BLADE P/N 76101-05101-041 S/N A137-00708

REMARKS - PILOT OR INSPECTOR	MAINTENANCE WORK	SIGNATURE	DATE
11-7-91 C/W AD 84-06-02, ASB 76-65-35A INSPECT OK - INSPECTED EDGE OF SPAR # CBT-TC-91-1512 AND NO CRACKS FOUND BLADE T.T. 4364:48	INSPECT SPAR PLUG - FOUND SIKORSKY MSG A/C TT 4364:48	[redacted]	11-7-91
3-27-92 4497:47 A/C TT	C/W AD 84-06-02 INSP T/R BLADE ELLIPTICAL SPAR PLUG.	[redacted]	3-27-92
11-20-92 4788:20	C/W AD 84-06-02 INSP. OF T/R BLADE ELLIPTICAL SPAR PLUG	[redacted]	11-20-92
7-25-93 TT: 5078:49	C/W ASB 76-65-43 INSP. Bumper pads For proper gaps RED - I.B .107 Yellow I.B .105 O.B .100 O.B .93	[redacted]	7-25-93
10-12-93 TT 5229:22	C/W ASB 76-65-44 T/R BLADE FAIRING MODIFICATION	[redacted]	10-12-93
	C/W AD 84-06-02 ELLIPTICAL PLUG INSPECTION	[redacted]	10-12-93
6-1-94 TT: 5721:05	C/W AD 84-06-02 Inspection of T/R Blade Elliptical Spar Plug.	[redacted]	6-2-94

SA 7343.3

SIKORSKY AIRCRAFT Division of UNITED TECHNOLOGIES

AIRCRAFT MAINTENANCE LOG

REMARKS - PILOT OR INSPECTOR	MAINTENANCE WORK	SIGNATURE	DATE
8-27-94	I/A/W AD94-14-20 BY INSTALLATION OF 76102-05004-111 RETAINING PAD I/A/W 76-65-35A PARA 2C. AD84-06-02 NOW doesnt APPLY	[redacted]	8-27-94
8-30-96	c/w insp of T/R blades I/A/W Zone 3, item 7 - Repainted T/R blade assy at comp TT 7028.31	[redacted]	[redacted]
11-19-98 total time (ACFT. 7712.5)	c/w AD 94-14-20 T/R plug insp. [redacted]	[redacted]	11-19-98
8-10-01 X-rayed, ultrasonic inspected per QATI 3016 and 3098, performed 1500 hour inspections, inspected pitch horns per AD 89-07-12, complied with ASB 76-65-41A 76-65-43 & 76-65-46; Replaced both supplemental polyurethane strips per SA 4047-76-5 section 4-16 rev. 9; Replaced both primary polyurethane strips per RS-018M-1; Replaced pivot retainers 4 ea per RS-22L-V; Replaced pivot bearings 4 ea per MM SA 4047-76-2 section 65-21-01; Repaired skin on paddle A per RS-22M-VII; Repaired wire mesh and performed 4000 hr torque check per RS-018-III and RS-002N-III; Replaced tips on both paddles per RS-021C-II; Replaced and reassembled center hardware per MM SA 4047-76-2 section 65-21-01; Refinished paint per Drwg. No. 76101-05020; Static balanced per drwg. 76101-11528 admt. 20 and drwg. 76101-11529 rev. C @ TTSN 7905.1 Pertinent details pertaining to the inspection, alteration and/or repairs are on file at this repair station under W.O. 29506 FAA CRS RW1R467K.		[redacted]	8-10-01
10/11/01 575CH ACTT 1083.3	C/W AD94-14-20 & ASB76-65-35B By Arrow Aviation - WO - 01-177		
12-30-03 X-rayed, ultrasonic inspected per QATI 3016 and 3098, performed all scheduled inspections up to and including 1500 hr. Inspection in accordance with Composite Materials Manual 4047-76-5 table 2-2 and Maintenance Manual 4047-76-2, 65-21-00 and 65-21-01, inspected elliptical plug for debonding per A.D. 94-14-20, complied with ASB 76-65-43 & 76-65-46, Replaced spar plug per RS-0221-II; Repaired trailing edge damage on paddle A per RS-018M-IV; Repaired gouge in pitch horn bushing area on both paddles per ORI 76100-018 part VI; Replaced both primary polyurethane strips per RS-018M-1; Replaced both supplemental polyurethane strips per SA 4047-76-5 section 4-16 rev. 9; Replaced and reassembled center hardware per MM SA 4047-76-2 section 65-21-01; Refinished paint per Drwg. No. 76101-05020; Static balanced per drwg 76101-11528 admt. 20 and drwg 76101-11529 rev. C @ TTSN 9663.9 All repairs were performed utilizing designed, process specification, drawings, manuals or other media supplied by the manufacture. Pertinent details to the inspection, alteration and repairs are on file at this repair station under W.O. 32570 FAA CRS RW1R467K.		[redacted]	12-30-03

UNITED TECHNOLOGIES SIKORSKY AIRCRAFT

COMPONENT LOG CARD

PART NO. 76101-05101-041 NOMENCLATURE Tail Rotor Blade SERIAL NO. A137-00708 X

AIRCRAFT NO.	DATE INSTALLED	DATE REMOVED	AIRCRAFT TIME	COMPONENT TIME T.S.O.	COMPONENT TIME T.T.	REASON FOR REMOVAL

OVERHAUL INTERVAL

DATE	REPAIRS OR OVERHAULS	SIGNATURE	CERTIFICATE NO.
	10-25-07 W/O 36671 X-rayed per MIL-STD-453, ultrasonic inspected per QATI 3098, 3261 & 3291, performed all scheduled inspections up to and including 1500 hr. inspection in accordance with composite MM 4047-76-5 table 3-2 and MM 4047-76-2, 65-21-00 and 65-21-01, complied with AD 94-14-20 R-1 & 89-07-12, ASB 76-65-40A, 76-65-41, 76-65-43B, 76-65-46 & 76-65-60; Performed 4000 hour torque check per RS-022N-III; Replaced both pitch horns per RS-040C-I; Replaced both primary poly strips per RS-018M-1; Replaced both secondary poly strips per SA 4047-76-5 section 4-16; Replaced laminar pad on spar of paddle B on outboard & pylon side per RS-022L-IV; Repaired skin damage on paddle B pylon side at sta. 9.0 to sta. 10.25 per OREI 2007-29; Replaced and reassembled center hardware, 2 ea fairings, 4 ea pivot bearings & 4 ea spandwise brackets per MM SA 4047-76-2 section 65-21-01; Refinished paint per drwg. 76101-05020; Static balanced per drwg. 76101-11528 amdt. 20 and drwg. 76101-11529 @ T.T. 12222.3	[redacted]	[redacted]
MAR/22/2012	I Inspected I.A.W. CMM-SA-4047-76-5 Table 2-2, Installed boot on "A" & "B" paddles per MM-SA-4047-76-2 (65-21-01) page 203, Replaced polyurethane supplemental strips on "A" & "B" paddles per CMM-SA-4047-76-5 proc. 10 (4-16), Complied with ASB 76-65-60 (spar inspection 500 hour), Complied with 1500 hr inspection per SA-4047-76-2-1 checkl zone 3, item 4-4A, Refinished per CMM-SA-4047-76-5 proc. 6 (4-10), Balanced tail rotor component. All actions performed per Sikorsky and BHTI specs. Reference W.O. # S76T-137-00708. T.T. 14800.5	[redacted]	[redacted]

SIKORSKY AIRCRAFT UNITED TECHNOLOGIES ... HELICOPTERS 884750 (ERA 58C)

AUG 13 2001

RECEIVING INSP.

COMPONENT CARD A137-00708X

PART NO. 76101-05101-041 NOMENCLATURE TAIL ROTOR BLADE SERIAL NO. A137-00708 X

AIRCRAFT NO.	DATE INSTALLED	DATE REMOVED	AIRCRAFT TIME	COMPONENT TIME T.S.O.	COMPONENT TIME T.T.	REASON FOR REMOVAL
760274	30Nov.1984		NEW	NEW	NEW	
760274		2 OCT 87	2230.0	New	2230.0	C/W 76-65-40
760274	13 Nov 87		2315.9	New	2230.0	
760274		17 NOV 90	3915.0	NEW	3915.0	REPAINT
~~760274~~	29 NOV 90		3915.0		3915.0	
760274		11-17-90	3915.0		3829.1	repaint
N579EH 760274	11-29-90		3915.0		3829.1	
N579EH 760274		6-22-01	7991.0		7905.1	erosion
575EH	10-9-01		1082.3		7905.1	
575EH	—	10-2-03	3341	~~[illegible]~~	9166.9	Erosion
N578EH	6-16-04	—	5669.4	—	9663.9	
578EH	—	2/10/07	8227.8	—	12222.3	Due Removal [redacted]
578EH	11-24-07	—	9327.3		12222.3	
578EH	—	1-7-2012	11905.5	tsi-312.1	14800.5	paint errosion; 1500hr [redacted]

[redacted]

ERA HELICOPTERS
JAN 05 2004
RECEIVING INSP.

SA 7343-9 REV A.

OVERHAUL INTERVAL

RETIREMENT TIME

SIKORSKY AIRCRAFT — Division of UNITED TECHNOLOGIES

COMPONENT MAINTENANCE RECOR

DATE	REPAIRS OR OVERHAULS	SIGNATURE	CERTIFICATE NO.
	Mfg. Date of Tail Rotor Blade #1: 6-14-84 #2: 6-14-84		
	Mfg. Date of Spar: 5-16-84		
4September1984	A.D. 84-06-02, Amd. 39-4829 - Inspection of Elliptical Centering Plug and Installation of Retaining Pad (A.S.B. 76-65-35A)	[redacted]	[redacted]
3 FEB87 ~~3 MAR 84~~	AD 86-19-14 ASB 76-65-38 INSPECTION OF T/R BLADE; REMOVAL of blade from SERVICE by Ser. Number	[redacted]	[redacted]
SEPT 15, 1986	ASB 76-65-37 REINSPECTION OF T/R ~~SPAR~~ BLADE ASSY N/A by ~~Serial Number~~ COMPONENT TIME	[redacted]	[redacted]
OCT 28 1986	ASB 76-65-39 INSPECT T/R bladespar for disbonding.	[redacted]	[redacted]
DEC 19 1	BLADE SPAR MUST BE INSPECTED PER S/B ASB 76-65-40 AND FOUND TO MEET BL MIN. FACTORS INSPECTION CRITERIA	[redacted]	[redacted]
4-25-88	C/W AD 84-06-02 / ASB 76-65-35A INSP SPAR ELIPTICAL PLUG	[redacted]	[redacted]
10/24/88	C/W AD 84-06-02 INSTALL SPAR ELIPTICAL PLUG	[redacted]	[redacted]
12/29/88	ASB 76-65-35A Tail Blade ELIPTICAL Plug Insp		
11/17/90	SANDED + REPAINTED BLADES I/A/W SIKORSKY INSTRUCT		
11/17/90	C/W 500HR INSP I/A/W COMPOSIT MANUAL INCLUDING		
11/17/90	AD 84-06-02 T/R SPAR PLUG INSP C/W		